

UNCLASSIFIED

AD NUMBER
AD892535
NEW LIMITATION CHANGE
TO Approved for public release, distribution unlimited
FROM Distribution authorized to U.S. Gov't. agencies only; Test and Evaluation; 23 JUN 1971. Other requests shall be referred to Air Force Flight Dynamics Laboratory, Attn: FY, Wright-Patterson AFB, OH 45433.
AUTHORITY
AFWAL ltr, 13 Mar 1981

THIS PAGE IS UNCLASSIFIED

THIS REPORT HAS BEEN DELIMITED
AND CLEARED FOR PUBLIC RELEASE
UNDER DOD DIRECTIVE 5200.20 AND
NO RESTRICTIONS ARE IMPOSED UPON
ITS USE AND DISCLOSURE.

DISTRIBUTION STATEMENT A

APPROVED FOR PUBLIC RELEASE;
DISTRIBUTION UNLIMITED.

AFFDL-TR-71-5
PART I, VOL II

DDC
2

**SUBSONIC UNSTEADY AERODYNAMICS
FOR GENERAL CONFIGURATIONS**

PART I, VOL II—COMPUTER PROGRAM H7WC

*J. P. GIESING
T. P. KALMAN
W. P. RODDEN*

DDC
RECEIVED
MAR 22 1972
B

TECHNICAL REPORT AFFDL-TR-71-5, PART I, VOL II

NOVEMBER 1971

Distribution limited to U. S. Government agencies only; test and evaluation; statement applied 23 June 1971. Other requests for this document must be referred to Air Force Flight Dynamics Laboratory, (FY), Wright-Patterson Air Force Base, Ohio.

**AIR FORCE FLIGHT DYNAMICS LABORATORY
AIR FORCE SYSTEMS COMMAND
WRIGHT-PATTERSON AIR FORCE BASE, OHIO**

AD 892535

AD No. _____
DDC FILE COPY

155

NOTICE

When Government drawings, specifications, or other data are used for any purpose other than in connection with a definitely related Government procurement operation, the United States Government thereby incurs no responsibility or any obligation whatsoever; and the fact that the Government may have formulated, furnished, or in any way supplied the said drawings, specifications, or other data, is not to be regarded by implication or otherwise as in any manner licensing the holder or any other person or corporation, or conveying any rights or permission to manufacture, use, or sell any patented invention that may in any way be related thereto.

REVISION No	
CFST	WHITE SECTION <input type="checkbox"/>
DOC	DIFF SECTION <input checked="" type="checkbox"/>
UNANNOUNCED	<input type="checkbox"/>
JUSTIFICATION	
BY	
DISTRIBUTION/AVAILABILITY CODES	
DIST.	AVAIL. and/or SPECIAL
B	

Copies of this report should not be returned unless return is required by security considerations, contractual obligations, or notice on a specific document.

AD 57-103 ✓

SUBSONIC UNSTEADY AERODYNAMICS FOR GENERAL CONFIGURATIONS

PART I, VOL II—COMPUTER PROGRAM H7WC

*J. P. GIESING
T. P. KALMAN
W. P. RODDEN*

Distribution limited to U. S. Government agencies only; test and evaluation; statement applied 23 June 1971. Other requests for this document must be referred to Air Force Flight Dynamics Laboratory, (FY), Wright-Patterson Air Force Base, Ohio.

FOREWORD

This report was prepared by the Douglas Aircraft Company, Long Beach, California, for the Aerospace Dynamics Branch, Vehicle Dynamics Division, Air Force Flight Dynamics Laboratory, Wright-Patterson Air Force Base, Ohio, under Contract F33615-70-C-1167. This research was conducted under Project 1370, "Dynamic Problems in Military Flight Vehicles," and Task 137003, "Prevention of Dynamic Aeroelastic Instabilities in Advanced Military Aircraft." Mr. S. J. Pollock of the Aerospace Dynamics Branch was Task Engineer.

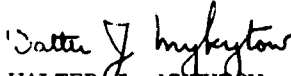
This report consists of two parts with two volumes for each part. This volume, Volume II of Part I is the Computer Program H7WC. Volume I of Part I consists of the method of direct application of nonplanar lifting surface elements. Volume I of Part II, contains a method which uses an image system and an axial singularity system to account for the effects of the bodies and Volume II of Part II is the Computer Program N5KA.

The work reported herein was conducted during the period of December 1969 to September 1970.

The Principal Investigator was Joseph P. Giesing. Mrs. Terez P. Kalman was responsible for implementing the method on the computer. Donald H. Larson aided in this implementation and acted as consultant for computer problems. Dr. William P. Rodden, a McDonnell Douglas Company Consultant, contributed many valuable ideas to the project. Others have made significant contributions to this project including Messrs. D. S. Warren and W. E. Henry.

The report was submitted by the authors in November 1970 for publication as an AFFDL Technical Report.

This technical report has been reviewed and is approved.


WALTER J. MYKYTOW
Asst. for Research & Technology
Vehicle Dynamics Division

A B S T R A C T

Two methods of accounting for body-lifting surface interference in unsteady flow are considered. The first method is described in Part I of this report, while the second will be described in Part II to follow.

The first method is a direct application of nonplanar lifting surface elements to both the lifting surfaces and the body surfaces. The body is treated as an annular wing. This type of idealization must be used with an axial doublet introduced to account for body incidence effects. The undesirable effects of the annular wing representation are then reduced.

The second approach, to be described in Part II, uses an image system and an axial singularity system to account for the effects of the bodies.

This volume contains the computer program and contains the FORTRAN listing. Volume I contains the development of the theory, correlation of theory with experimental data, and the parametric study.

CONTENTS

	Page
1.0 Introduction	1
2.0 Computer Program H7WC	5
2.1 Input Procedure	5
2.1.1 Input Sheets	5
2.1.2 Description of Input Data	9
2.2 Example Case	19
2.2.1 Input Sheets	20
2.2.2 Output	26
2.3 Blank and Labeled Common Blocks	38
2.3.1 Blank Common Blocks	38
2.3.2 Labeled Common Blocks	42
2.4 Logical Tape Units	47
2.5 Description of Subroutines	48
2.5.1 MAIN	48
2.5.2 PART1	50
2.5.3 GENB	52
2.5.4 GENW	54
2.5.5 GENQ	57
2.5.6 AUGW	60
2.5.7 TKER	63
2.5.8 PART2	64
2.5.9 INCRO	65

CONTENTS (CONT ' D)

	Page
2.5.10 KERNEL	68
2.5.11 IDF1	69
2.5.12 IDF2	70
2.5.13 SNPDF	71
2.5.14 SSLECX	73
2.5.15 CXSS	75
2.5.16 SOLVIT	76
2.5.17 QUAS	78
2.5.18 FUTSOL	80
2.5.19 AIC	81
2.5.20 AERO	83
2.5.21 FINAL	85
2.5.22 GENF	87
2.6 Block Diagram	89
2.7 Program Listing	94

NOMENCLATURE

A	Reference total area
[A]	Matrix of box areas
a_{inm}	Polynomial mode coefficients for mode i
AIC	Matrix of influence coefficients relating generalized forces to generalized deflections for submodes
c	Chord length
\bar{c}	Reference chord length
C_m	Pitching moment coefficient $\left(\frac{\text{moment}}{q A \bar{c}} \right)$, (+ nose up)
C_n	Yawing moment coefficient $\left(\frac{\text{moment}}{q A s} \right)$ (+ nose right)
C_l	Rolling moment coefficient $\left(\frac{\text{moment}}{q A s} \right)$ (+ clockwise)
C_Y	Force coefficient in y-direction $\left(\frac{\text{Force}}{q A} \right)$ (+ out right wing)
C_Z	Force coefficient in z-direction $\left(\frac{\text{Force}}{q A} \right)$ (+ vertically up)
c_m	Local moment coefficient $\left(\frac{\text{moment}}{q c^2} \right)$
c_n	Local normal force coefficient $\left(\frac{\text{normal force}}{qc} \right)$
e	Strip semi-width
ϵ	Normalized deflection normal to surface (h/s)
h	Deflection normal to surface
K	Kernel
k_r	Reduced frequency, $\omega \bar{c} / 2U_\infty$
L	Lift
M	Mach number; also moment
Q_{ij}	Generalized force
q	Dynamic pressure
\bar{q}	Generalized coordinates
R_0	Body radius
r	$\sqrt{(y-\eta)^2 + (z-\xi)^2}$

s	Semi-span
U_∞	Free-stream velocity
W	Unnormalized normalwash
w	W/U_∞
w_r	$w - w_b$
x, y, z	Coordinates of receiving points
$\bar{x}, \bar{y}, \bar{z}$	Element coordinates of receiving points
x_0	Gust reference axis
α	Angle of attack; also a function defined by Equation 2.1-15
β	$\sqrt{1 - M^2}$; also control surface deflection
Γ_g	Gust dihedral angle. ($\Gamma_g = 0$ if gust velocity is vertical)
γ	Dihedral angle
ΔC_p	Lifting pressure coefficient $\frac{P_{\text{lower}} - P_{\text{upper}}}{q}$
δ	Symmetry index (right and left symmetry); also tab deflection
ϵ	Ground effect index
λ	Wave length
ξ, η, ζ	Coordinates of sending points
$\bar{\xi}, \bar{\eta}, \bar{\zeta}$	Coordinates of sending points in element coordinates
ρ_0	Density at sea level
σ	Lateral coordinate in the plane of the surface
ω	Frequency

Subscripts and Superscripts

a	body axis
B	body
c	center
f	Body or fuselage
g	gust

i	Deflection mode
J	Pressure mode
L. E.	Leading edge
LL	Lower left
LR	Lower right
R	Axis about which moments are taken
r	Receiving
s	Sending
UL	Upper left
UR	Upper right
y	y-direction
z	z-direction
1/4	One quarter chord point of a lifting surface box
3/4	Three quarters chord point of a lifting surface box

1.0 INTRODUCTION

Computer program H7WC is the result of modifying and extending program H7WB. Program H7WB generates aerodynamic influence coefficients and gust loads for nonplanar surfaces. The modifications to this program include:

- (1) An extension to accept polynomial mode input. (Subroutine GENQ)
- (2) The addition of a capability to determine generalized forces. (Subroutine GENF)
- (3) The addition of a capability to account for body angle of attack and camber effects using an axial pressure doublet system. (Subroutines AUGW, TKER)
- (4) A refined version of the Doublet-Lattice Method to properly handle problems involving small vertical gaps between wing and tail surfaces. (Subroutine IDF2)
- (5) The addition of the Quasi-Inverse for solving large numbers of modes or submodes, and for solving future modes with a minimum of computing effort. (Subroutines QUAS, FUTSOL, CXSS)

Program H7WC can handle almost any type of configuration if it is idealized properly. The axial pressure doublets give the proper slender body flow field for an axially symmetric body of any shape. The body/lifting-surface interference is taken care of using lifting surface panels on the body surface. The interference elements must lie on an idealized body of constant cross sectional shape. The rules that apply to the placement of a grid of boxes on the aircraft lifting surfaces also apply to the interference surfaces on the bodies. These rules are:

- (1) Boxes (trapezoidal elements) are arranged in strips parallel to the free-stream. The aspect ratio of such boxes should not be large. For the unsteady case, an aspect ratio of order unity or less is preferred.

- (2) Surface intersections, surface edges, fold lines, and hinge lines should lie on box boundaries.
- (3) For wing-tail type configurations the strip boundaries on the tail must be aligned with those on the wing. Strips should be concentrated near the wing tips, near control surface edges and in any region where the span loading changes rapidly.
- (4) The box length must be small compared to the basic wave length.

$$\frac{\Delta x}{\lambda} \leq 0.04$$

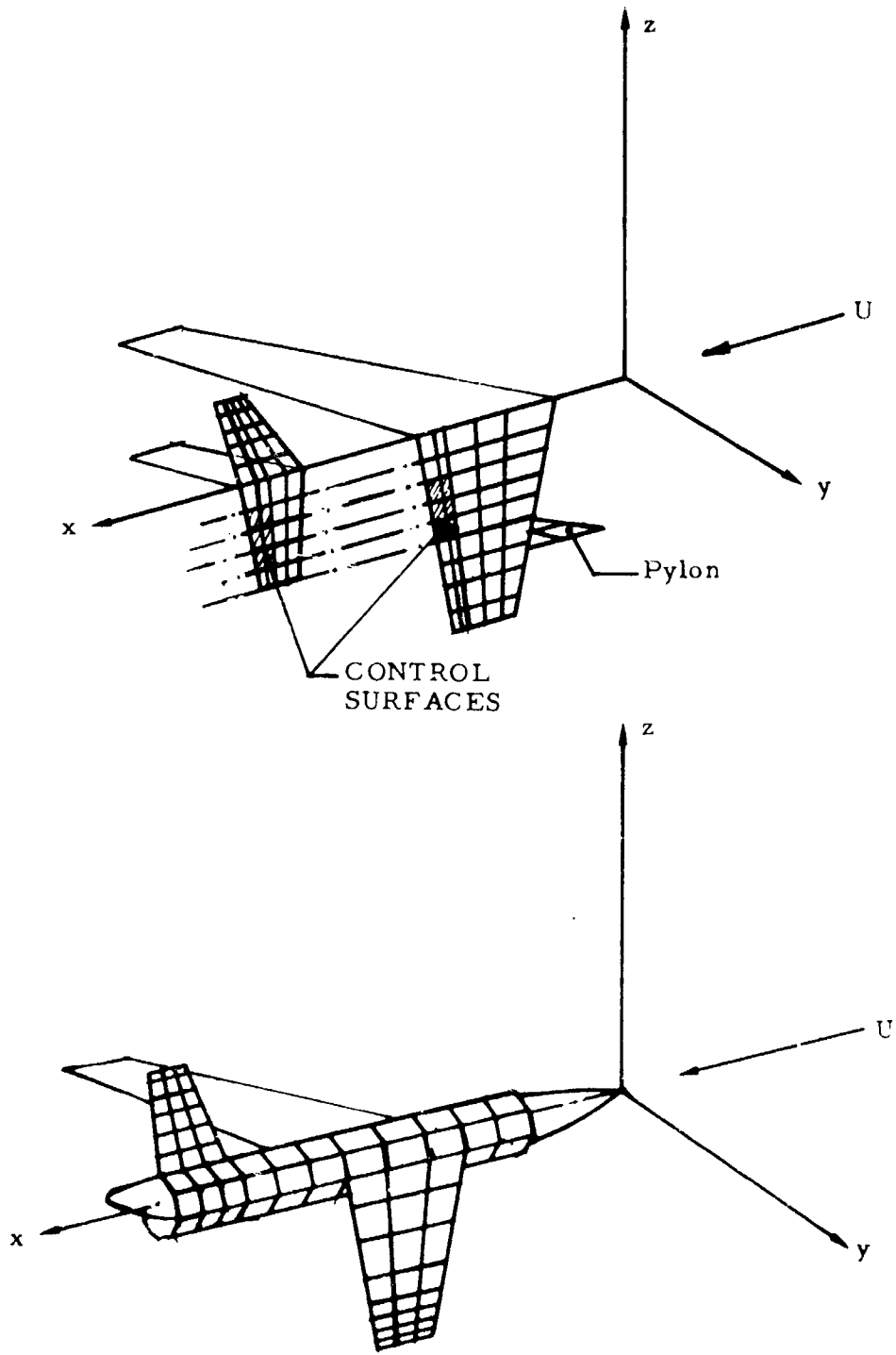
Boxes should also be concentrated near control surface hinge-lines and in all regions where the upwash boundary conditions are discontinuous.

- (5) The body cross-sectional shape on which interference lifting surfaces are placed must be constant in the x-direction even for wing-tail problems. (This does not apply to the slender body solution for which the radius distribution is used.)

Sketch 1,1-1 gives examples of these 5 rules.

Because of the numerical techniques involved, the two-dimensional radial distance, $\sqrt{(y-\eta_e)^2 + (z-\zeta_e)^2}$, between a control point, y, z , (3/4-chord point of a box) and a box edge, η_e, ζ_e , may never approach zero. It does not matter whether the control point is on the same or on a different surface or whether it is up-or downstream from the box in question. This is easily understood for a control point in the wake of the box since the control point may not approach the bound leg of the horseshoe vortex. However, it is also true upstream due to the approach taken for the unsteady effects. (If $k_r = 0$ the upstream effect at $\sqrt{(y-\eta)^2 + (z-\zeta)^2} \rightarrow 0$ vanishes).

Wing-body-tail problems must be handled using only one interference cross-section. That is, one constant cross-sectional shape must be used both near the wing and near the tail. If this rule is not followed the wakes of the lifting surface element lying on the cross-section near the wing may impinge on the tail surface.



Sketch 1.1-1

When a body mode is input it must be input both to the axial elements and to the interference body panels. It must be remembered, however, that the body is in motion either in the z-or y-directions while the lifting surfaces on the body are in motion normal to themselves. Thus the mode input for a particular body interference panel must be multiplied by $\cos\gamma$ for (+) z-motion and $-\sin\gamma$ for (+) y-motion where γ is the dihedral angle of the panel.

The following list gives the program limits.

1. The maximum number of unknowns, i. e., the aerodynamic boxes on the lifting surfaces plus the number of body elements, is 400.
2. The maximum number of aerodynamic degrees of freedom, in case of AIC calculations, is 130.
3. The maximum number of degrees of freedom per strip in case of AIC calculations is restricted to 4.*
4. The maximum number of modal right-hand sides is 40 per case.
5. The maximum number of aerodynamic boxes per spanwise division (strip) is 30.
6. The maximum number of spanwise divisions (strips) is 50.
7. The maximum number of bodies is 20.

Program H7WC was used successfully on the following computers: GE635, IBM 7094 and IBM 360/65. Core requirement for the program on the various computers is as follows:

<u>GE635</u>	<u>IBM 7094</u>	<u>IBM 360/65</u>	<u>CDC 6600</u>
28K words	32K words	126K bytes	100K bytes

If greater core capacity is available it is desirable to use it up to 260K bytes. The CDC version is slightly modified in dimensions.

* This restriction comes from operational requirements on the IBM 7094. The number of degrees of freedom can easily be increased to the 7 discussed in Appendix D in Part I, Vol 1 by increasing the dimension 4 to 7 in subroutines AIC, AERØ, and FINAL.

2.0 COMPUTER PROGRAM H7WC

2.1 Input Procedure

2.1.1 Input Sheets

The input sheets for program H7WC are shown on the next three pages. The first three cards represent general case data that is input once per case. The next four* cards (4, 5, 6, 7) represent panel data that is repeated per panel. The panel data must be input in a certain order. The panel data for all "regular" lifting surfaces, (wing, tail, etc.) are input first. The panel data for all interference lifting surfaces (used for the bodies) is input last.

If a body can oscillate in both the z- and y-directions (nacelle) then it must be replaced by two bodies: one that moves in the z-direction and one that moves in the y-direction. The subsequent three** cards (8, 9, 10) are body data necessary for the slender body analysis. This data is repeated per body. Card 11 is used for miscellaneous modal and AIC data and must be completed in either case. Card 12 is used when a gust is desired. Card 13 indicates a series of cards that are used to identify lifting surface boxes with strips. Card 14 indicates one or more cards that identify the degrees of freedom to be used for each panel. This card is necessary only if AIC's are desired.

Cards 15 and 16 are used to identify axes for the AIC degrees of freedom. Card 17 is used for modal control data. Card 18 is used to input control data for panel modes, while card 19 represents a series of cards, repeated as necessary, to indicate polynomial mode coefficients for panels. Cards 20 and 21 are similar to cards 18, 19 except the modes pertain to the z and y modes of the bodies.

* There may be more than four cards depending on the number necessary to present all θ_i and r_i values.

** There may be more than three cards depending on the number necessary to present F and RAD.

ENGINEER _____

PHONE _____

DATE _____

53 54 55 56 57 58
PUNCH IN ALL CARDS

73 74 75 76
H 7 W C
PROGRAM NO.

1	2	3	4	5	6	7	8	9	10	11	12	13	14	15	16	17	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32	33	34	35	36	37	38	39	40	41	42	43	44	45	46	47	48	49	50	51	52	53	54	55	56	57	58	59	60																
HEADER																																																																											
M	A															C															KR ₁															ε NP NB NR NC Ø RE																													
X1	X2															X3															X4															Y1															Y2														
Z1	Z2															NS NC															CØEF															CØEF															MK1 MK2														
Zc	Yc															NE NZ															I, NS															I, NS															MK1 MK2														
	F _k															F _k															k = 1, NF															k = 1, NF																													
	RAD _k															RAD _k															k = 1, NF															k = 1, NF																													

DO NOT PUNCH BLANK COLUMNS
NO UNDERPUNCHES IN SIGN FIELDS.

REPEAT PER PANEL REPEAT PER BODY

SEQ. NO.	77	78	79	80
	01	02	03	04
	05	06	07	08
	09	10		

ENGINEER

PHONE

DATE

53 54 55 56 57 58

73 74 75 76
H 7 W C

PUNCH IN ALL CARDS

PROGRAM NO.

SEQ NO.	1	2	3	4	5	6	7	8	9	10	11	12	13	14	15	16	17	18	19	20	21
77787980	NSRQNB1	NB2	NPR1	NPR2	NPR3	JØBNØ	VEL	WG	JØBNØ2	NCUS	JSP	NPC	NSV	NBV	NY						
11	GZRØ	XZRØ																			
12																					
13	LIM1	LIM2	LIM3			LIM															
14						IS															
15						XHI															
16						XHØ															
17	NMNT	N5	N6	N7		NTPM															
18	NR	NCNA	N8			NTP1															
19	N	M				NR	NO	NA	NB												
20	NR	NO	NA	NB		ARQ															
21	N	M				ARQ															

REPEAT PER PANEL

PANEL MODES
BODY MODES

DO NOT PUNCH BLANK COLUMNS
NO UNDERPUNCHES IN SIGN FIELDS.

DIRECTIONS FOR KEYPUNCH

ENGINEER _____

PHONE _____

DATE _____

53 64 65 66 67 68

73 74 75 76
H 7 W C

PROGRAM NO.

CASE

PUNCH IN ALL CARDS

1	2	3	4	5	6	7	8	9	10	11	12	13	14	15	16	17	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32	33	34	35	36	37	38	39	40	41	42	43	44	45	46	47	48	49	50	51	52	53	54	55	56	57	58	59	60	61	62	63	64	65	66	67	68	69	70	71	72	73	74	75	76	77	78	79	80	81	82	83	84	85	86	87	88	89	90	91	92	93	94	95	96	97	98	99	100	101	102	103	104	105	106	107	108	109	110	111	112	113	114	115	116	117	118	119	120	121	122	123	124	125	126	127	128	129	130	131	132	133	134	135	136	137	138	139	140	141	142	143	144	145	146	147	148	149	150	151	152	153	154	155	156	157	158	159	160	161	162	163	164	165	166	167	168	169	170	171	172	173	174	175	176	177	178	179	180	181	182	183	184	185	186	187	188	189	190	191	192	193	194	195	196	197	198	199	200	201	202	203	204	205	206	207	208	209	210	211	212	213	214	215	216	217	218	219	220	221	222	223	224	225	226	227	228	229	230	231	232	233	234	235	236	237	238	239	240	241	242	243	244	245	246	247	248	249	250	251	252	253	254	255	256	257	258	259	260	261	262	263	264	265	266	267	268	269	270	271	272	273	274	275	276	277	278	279	280	281	282	283	284	285	286	287	288	289	290	291	292	293	294	295	296	297	298	299	300	301	302	303	304	305	306	307	308	309	310	311	312	313	314	315	316	317	318	319	320	321	322	323	324	325	326	327	328	329	330	331	332	333	334	335	336	337	338	339	340	341	342	343	344	345	346	347	348	349	350	351	352	353	354	355	356	357	358	359	360	361	362	363	364	365	366	367	368	369	370	371	372	373	374	375	376	377	378	379	380	381	382	383	384	385	386	387	388	389	390	391	392	393	394	395	396	397	398	399	400	401	402	403	404	405	406	407	408	409	410	411	412	413	414	415	416	417	418	419	420	421	422	423	424	425	426	427	428	429	430	431	432	433	434	435	436	437	438	439	440	441	442	443	444	445	446	447	448	449	450	451	452	453	454	455	456	457	458	459	460	461	462	463	464	465	466	467	468	469	470	471	472	473	474	475	476	477	478	479	480	481	482	483	484	485	486	487	488	489	490	491	492	493	494	495	496	497	498	499	500	501	502	503	504	505	506	507	508	509	510	511	512	513	514	515	516	517	518	519	520	521	522	523	524	525	526	527	528	529	530	531	532	533	534	535	536	537	538	539	540	541	542	543	544	545	546	547	548	549	550	551	552	553	554	555	556	557	558	559	560	561	562	563	564	565	566	567	568	569	570	571	572	573	574	575	576	577	578	579	580	581	582	583	584	585	586	587	588	589	590	591	592	593	594	595	596	597	598	599	600	601	602	603	604	605	606	607	608	609	610	611	612	613	614	615	616	617	618	619	620	621	622	623	624	625	626	627	628	629	630	631	632	633	634	635	636	637	638	639	640	641	642	643	644	645	646	647	648	649	650	651	652	653	654	655	656	657	658	659	660	661	662	663	664	665	666	667	668	669	670	671	672	673	674	675	676	677	678	679	680	681	682	683	684	685	686	687	688	689	690	691	692	693	694	695	696	697	698	699	700	701	702	703	704	705	706	707	708	709	710	711	712	713	714	715	716	717	718	719	720	721	722	723	724	725	726	727	728	729	730	731	732	733	734	735	736	737	738	739	740	741	742	743	744	745	746	747	748	749	750	751	752	753	754	755	756	757	758	759	760	761	762	763	764	765	766	767	768	769	770	771	772	773	774	775	776	777	778	779	780	781	782	783	784	785	786	787	788	789	790	791	792	793	794	795	796	797	798	799	800	801	802	803	804	805	806	807	808	809	810	811	812	813	814	815	816	817	818	819	820	821	822	823	824	825	826	827	828	829	830	831	832	833	834	835	836	837	838	839	840	841	842	843	844	845	846	847	848	849	850	851	852	853	854	855	856	857	858	859	860	861	862	863	864	865	866	867	868	869	870	871	872	873	874	875	876	877	878	879	880	881	882	883	884	885	886	887	888	889	890	891	892	893	894	895	896	897	898	899	900	901	902	903	904	905	906	907	908	909	910	911	912	913	914	915	916	917	918	919	920	921	922	923	924	925	926	927	928	929	930	931	932	933	934	935	936	937	938	939	940	941	942	943	944	945	946	947	948	949	950	951	952	953	954	955	956	957	958	959	960	961	962	963	964	965	966	967	968	969	970	971	972	973	974	975	976	977	978	979	980	981	982	983	984	985	986	987	988	989	990	991	992	993	994	995	996	997	998	999	1000	1001	1002	1003	1004	1005	1006	1007	1008	1009	1010	1011	1012	1013	1014	1015	1016	1017	1018	1019	1020	1021	1022	1023	1024	1025	1026	1027	1028	1029	1030	1031	1032	1033	1034	1035	1036	1037	1038	1039	1040	1041	1042	1043	1044	1045	1046	1047	1048	1049	1050	1051	1052	1053	1054	1055	1056	1057	1058	1059	1060	1061	1062	1063	1064	1065	1066	1067	1068	1069	1070	1071	1072	1073	1074	1075	1076	1077	1078	1079	1080	1081	1082	1083	1084	1085	1086	1087	1088	1089	1090	1091	1092	1093	1094	1095	1096	1097	1098	1099	1100	1101	1102	1103	1104	1105	1106	1107	1108	1109	1110	1111	1112	1113	1114	1115	1116	1117	1118	1119	1120	1121	1122	1123	1124	1125	1126	1127	1128	1129	1130	1131	1132	1133	1134	1135	1136	1137	1138	1139	1140	1141	1142	1143	1144	1145	1146	1147	1148	1149	1150	1151	1152	1153	1154	1155	1156	1157	1158	1159	1160	1161	1162	1163	1164	1165	1166	1167	1168	1169	1170	1171	1172	1173	1174	1175	1176	1177	1178	1179	1180	1181	1182	1183	1184	1185	1186	1187	1188	1189	1190	1191	1192	1193	1194	1195	1196	1197	1198	1199	1200	1201	1202	1203	1204	1205	1206	1207	1208	1209	1210	1211	1212	1213	1214	1215	1216	1217	1218	1219	1220	1221	1222	1223	1224	1225	1226	1227	1228	1229	1230	1231	1232	1233	1234	1235	1236	1237	1238	1239	1240	1241	1242	1243	1244	1245	1246	1247	1248	1249	1250	1251	1252	1253	1254	1255	1256	1257	1258	1259	1260	1261	1262	1263	1264	1265	1266	1267	1268	1269	1270	1271	1272	1273	1274	1275	1276	1277	1278	1279	1280	1281	1282	1283	1284	1285	1286	1287	1288	1289	1290	1291	1292	1293	1294	1295	1296	1297	1298	1299	1300	1301	1302	1303	1304	1305	1306	1307	1308	1309	1310	1311	1312	1313	1314	1315	1316	1317	1318	1319	1320	1321	1322	1323	1324	1325	1326	1327	1328	1329	1330	1331	1332	1333	1334	1335	1336	1337	1338	1339	1340	1341	1342	1343	1344	1345	1346	1347	1348	1349	1350	1351	1352	1353	1354	1355	1356	1357	1358	1359	1360	1361	1362	1363	1364	1365	1366	1367	1368	1369	1370	1371	1372	1373	1374	1375	1376	1377	1378	1379	1380	1381	1382	1383	1384	1385	1386	1387	1388	1389	1390	1391	1392	1393	1394	1395	1396	1397	1398	1399	1400	1401	1402	1403	1404	1405	1406	1407	1408	1409	1410	1411	1412	1413	1414	1415	1416	1417	1418	1419	1420	1421	1422	1423	1424	1425	1426	1427	1428	1429	1430	1431	1432	1433	1434	1435	1436	1437	1438	1439	1440	1441	1442	1443	1444	1445	1446	1447	1448	1449	1450	1451	1452	1453	1454	1455	1456	1457	1458	14
---	---	---	---	---	---	---	---	---	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	-----	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	------	----

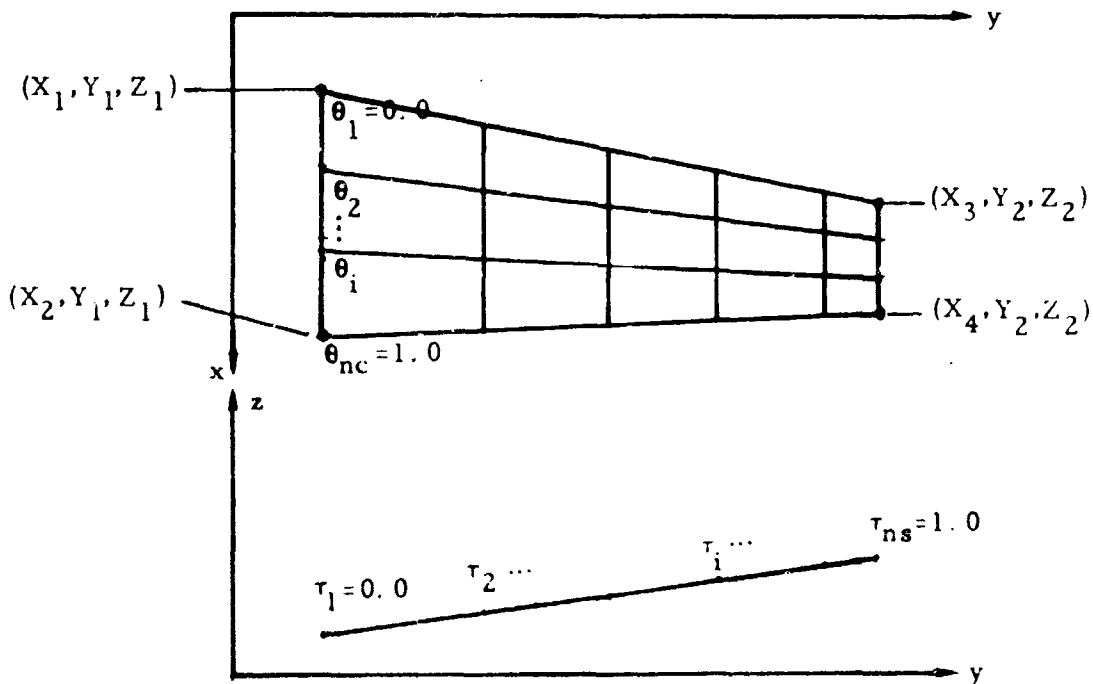
2. 1. 2 Description of Input Data

<u>Item No.</u>	<u>Description</u>
1	The header card is used to describe the current run being made. Any alphanumeric information can be used, placed in cc 1-60. The first 12 card columns should contain the tape title for the AIC's, in the case when AIC's are generated and are to be saved on master tape for later use.
2	<u>M, FMACH</u> - Mach number, usual definition.
3	<u>A, ACAP</u> - Reference area; usually total area of wing.
4	<u>c̄, REFCHD</u> - Reference chord, usually average chord of wing
5	<u>s, REFSPN</u> - Reference semispan.
6	<u>δ, NDEL</u> - Symmetry flag ($\delta = 1$ sym., $\delta = -1$, antisym., $\delta = 0$, no sym.).
7	<u>NP, NOPAN</u> - Total number of panels on all lifting surfaces.
8	<u>NB</u> - Total number of bodies.
9	<u>NRF</u> - Number of reduced frequencies for which the analysis has to be performed (repeated) - recommended maximum is 6 per case.
10	<u>NCORE</u> - Size of total solution, i. e., $NCORE = N \times M$ where N is the total number of elements or unknowns (including z and y fuselage elements) and M is the number of modes or the order of the AIC matrix if obtained.
11	<u>NI</u> - If "1", program assumes that case has been run previously and that a quasi-inverse exists on a tape (identified by NTPI) and only new modes are being run. If "0", program runs a new case.

Item No.

Description

- 12 N2 - If "0", polynomial modes are to be input. If "1" AIC's are to be calculated.
- 13 N3 - Data flag. If "1", matrix is printed; if "0", it is not printed.
- 14 N4 - Detail print flag for slender body solution; 1 to print, 0 otherwise.
- 15 KR - Array of reduced frequency, $KR = \frac{\omega \bar{c}}{2 U_\infty}$.
- 16, 17, 18 X1 X2 X3 - Panel edge coordinates.
- 19, 20, 21 X4 Y1 Y2 X1, Y1, Z1 (Inboard leading edge)
- 22, 23 Z1 Z2 X2, Y1, Z1 (Inboard trailing edge)
X3, Y2, Z2 (Outboard leading edge)
X4, Y2, Z2 (Outboard trailing edge)
- 24 NS - Number of spanwise divisions for panel considered.
- 25 NC - Number of chordwise divisions for panel considered.



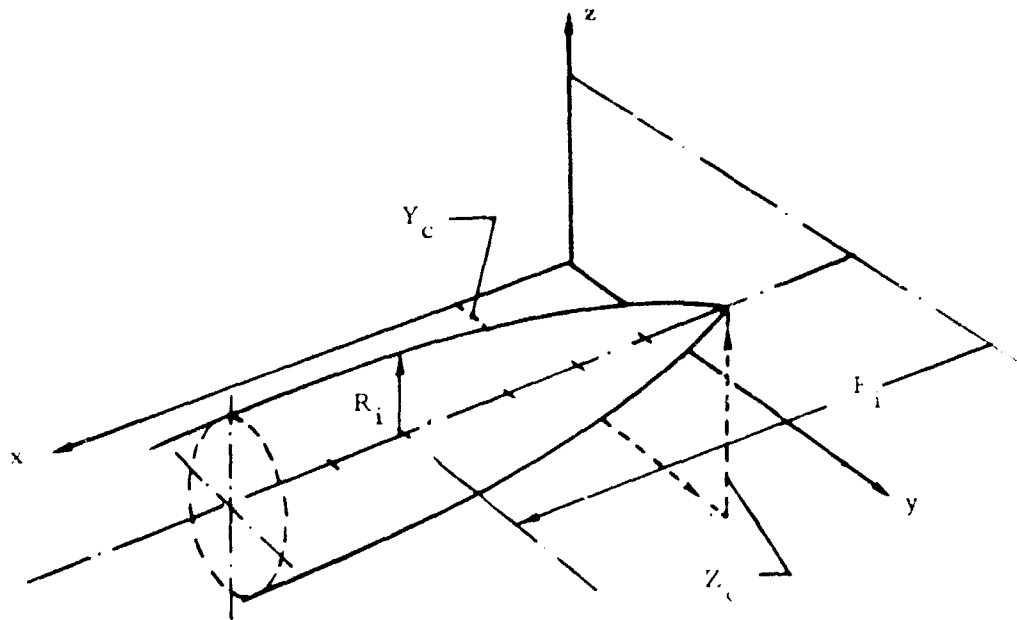
SKETCH 2.1.2-1

<u>Item No.</u>	<u>Description</u>
26	<u>COEF</u> - Scale factor for panel deflection modes. If no scale factor is desired, must be set to 1.0.
27	θ_i - Chordwise divisions in fraction of chord. Usually varies from 0 at leading edge to 1.0 at the trailing edge.
28	τ_i - Spanwise divisions in fraction of panel span. Usually 0.0 at inboard chord and 1.0 at outboard chord.

Repeat all panel data per panel (Items 16 through 28) Panels are input in the following sequence: wing, tail, etc., panels, then body interference panels.

29	Z_c - Z-coordinate of body axis.
30	Y_c - Y-coordinate of body axis.
31	<u>NF</u> - Number of divisions on body.
32	<u>NZ</u> - If "1", the program will allow for doublets in the Z-direction and thus will be able to account for Z-or upwash. If not, set NZ = 0.
33	<u>NY</u> - If "1", program will allow for Y-doublets and thus Y-or sidewash.
34	<u>COEF</u> - Scale factor for body deflection mode.
35	<u>MK1</u> - Each body has associated with it interference lifting surfaces. MK1 represents the first box on these surfaces.
36	<u>MK2</u> - The last box on the body interference lifting surface panels. Interference lifting surface panels are input after the regular lifting surfaces are input.
37	<u>E</u> - Endpoints of body divisions (starts at leading edge and ends at trailing edge). E x coordinates of body element end points.
38	<u>RAD</u> - Radius at endpoints.

Repeat all body data per body (Items 29 through 38)



SKETCH 2.1.2-2

<u>Item No.</u>	<u>Description</u>
39	<u>NSTRIP</u> - Total number of strips on all lifting surfaces.
40	<u>NB1</u> - Sequence number of first strip. Omit if no AIC's are desired.
41	<u>NB2</u> - Sequence number of last strip. Omit if no AIC's are desired.
42	<u>NPR1</u> - Print flag. If "1", print all AIC solutions. Otherwise use "0" (usually = 0).
43	<u>NPR2</u> - Print flag. If "1", print AIC's. Otherwise use "0" (usually = 0).
44	<u>NPR3</u> - Stability derivative flag. If "1", print static stability derivatives. If "3", print static-and dynamic stability derivatives. Otherwise use "0". Note that, if NPR3 = 3, NRF must be ≥ 2 and FR(1) must be 0.0. Usual setting is NPR3=1.
45	<u>JOBNO</u> - AIC tape identification. Ignore if no AIC's.
46	<u>JOBNO2</u> - Not used.
47	<u>NGUST</u> - Gust flag. Use "1" for gust. "0" otherwise.

<u>Item No.</u>	<u>Description</u>
48	<p><u>JSP</u> - Second symmetry plane (the plane $Z = 0$).</p> <p>-1 ground effect (antisymmetric)</p> <p>1 biplane or jet (symmetric)</p> <p>0 no symmetry</p>
49	<p><u>NPC</u> - Mode selector for AIC generation.</p> <p>Ignore if no AIC's. Use "0" if alternative #1 is used i. e. plunging, pitching, control surface rotation and tab rotation. Use "1" if alternative #2 is used, i. e., three cambering modes plus control surface and tab rotation.</p>
50	<p><u>NSV</u> - Total number of strips on all vertical panels which lie on the symmetry plane $Y = 0$.</p> <p>Vertical panels lying on $Y = 0$ must be input before the other panels. For example the vertical portion of the T-tail must be input first.</p>
51	<p><u>NBV</u> - Total number of boxes on vertical panels.</p>
52	<p><u>NYAW</u> - Stability derivative override flag. If "1", yaw coefficients will be calculated, if "0", pitch coefficients will be calculated.</p> <p>If $NDEL T = 1$, $NYAW = 0$</p> <p>$NDEL T = -1$, $NYAW = 1$</p> <p>$NDEL T = 0$, $NYAW = 0$ or 1</p>
53	<p><u>GZRO</u> - Gust reference plane dihedral in degrees. For 0 dihedral ($GZRO = 0$) the gust velocities are vertical (in the Z-direction). Omit if gust is not desired.</p>
54	<p><u>XZRO</u> - Gust reference point. Point at which sinusoidal gust velocity is unity. Omit if gust is not desired.</p>

<u>Item No.</u>	<u>Description</u>
55	<u>VEL</u> - Aircraft velocity in. /sec. Omit if gust is not desired.
56	<u>WG</u> - Gust vertical velocity at XZRO, in. /sec. Omit if gust is not desired.
Omit the gust card if the gust is not desired. (Items 53 through 56)	
57	<u>LIM</u> - Summation limits for chordwise integration. Usually LIM1 is the first box in a strip and LIM2 is the last box where all boxes are numbered consecutively.
58	<u>NOP</u> - Panel number.
59, 60, 61 62, 63, 64 65	IS1 IS2 IS3 - AIC degree-of-freedom selector. If alternative #1 is used (i. e. if NPC = 0) then IS1 through IS6 (IS7 not used) represent: plunging, pitching, control surface rotation, tab rotation, control surface plunging, tab plunging, respectively. If Alternative #2 is used then IS1 through IS7 represent: three cambering modes on the main surface, control surface rotation, tab rotation, control surface plunging, and tab plunging, respectively.
66	<u>XHI</u> - Reference axes measured in fractions of local chord on the inboard edge of a panel. (Further description following XHO.) Ignore if AIC's are not needed.
67	<u>XHO</u> - Reference axes measured in fractions of local chord on the outboard edge of a panel. For Alternative #1 5 quantities are needed: the elastic axis, the leading edge of the control surface, the rotation point of the control surface, the tab leading edge and rotation point, respectively. The data pair XHI and XHO are repeated for each panel.

Item No.Description

Omit data Items #58 through #67 if AIC's are not required.

- 68 NM, NMD - Total number of modes, i. e., max NQ.
- 69 NT, NTA - Total number of ARQ values.
- 70 N5 - If "1", save the quasi-inverse on tape so that future modes may be considered. Otherwise set to "0".
- 71 N6 - Type of mode input flag
 0 - polynomial
 1 - input h, dh/dx from tape
 2 - input h, dh/dx from cards
- 72 N7 - If "1", calculate pressure forces and moments. Use "0" otherwise.
- 73 NTPI - Quasi Inverse tape identification. Written on tape when Quasi Inverse is formed and used to identify the Quasi Inverse when it is retrieved for additional mode cases.
- 74 NTPM - Identification of input mode (h, dh/dx) tape.
-

The input to follow is associated with the input modes. Equations will be written to define the variables. For panels (lifting surfaces) the deflections will be normal to the surface while for bodies the deflections will be in the Z- (if any) and Y- (if any) directions. For the "NR"th panel or body the deflections in the "NQ"th mode are calculated as follows:

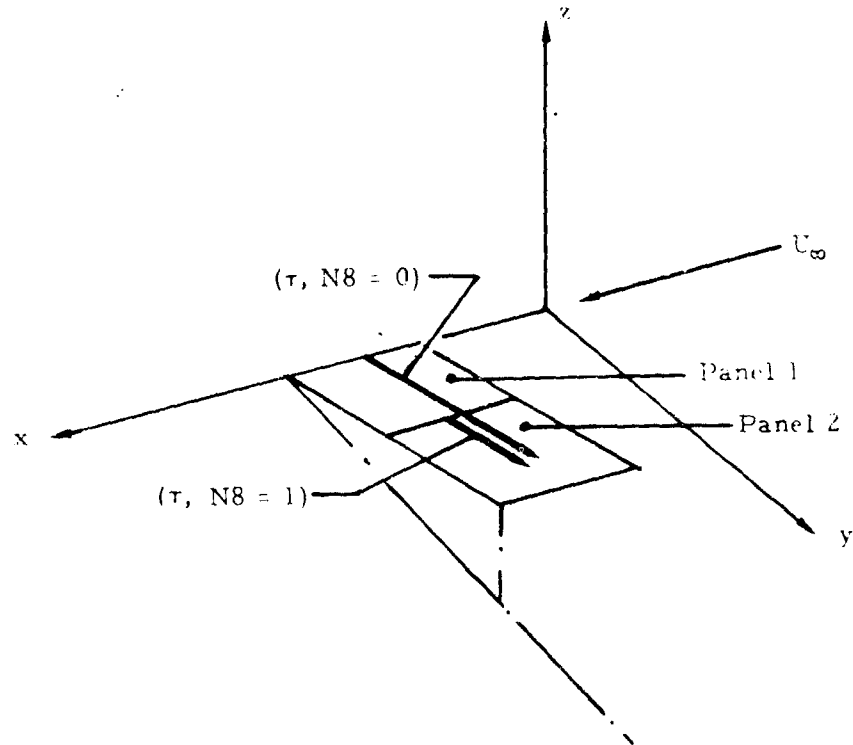
$$(f) \begin{matrix} (NR) \\ NQ \end{matrix} = \text{COEF}^{(NR)} \sum_{N=0,1,\dots} \sum_{M=0,1,\dots} \left(\frac{x}{s}\right)^N \left(\frac{r}{s}\right)^M \text{ARQ}_{(NR, NQ, N, M)}$$

where r is in a radial direction. The origin of the radius is either at the

origin of coordinates, when N8 = 0, or at the inboard edge (for the panel) or the axis (for a body) if N8 is set to 1.*

$$\tau = \sqrt{(y - (N8)Y_1^{(NR)})^2 + (z - (N8)Z_1^{(NR)})^2} \quad \text{for panels}$$

$$\tau = \sqrt{(y - (N8)Y_c^{(NR)})^2 + (z - (N8)Z_c^{(NR)})^2} \quad \text{for bodies}$$



SKETCH 2.1.2-3

* Since each panel is planar, τ is a spanwise distance in the plane measured from its inboard edge or from the origin $y = z = 0$; τ is a radial distance only in the sense that each panel may have a different dihedral. The use of this radial distance is not meant to imply that the panels are curved.

<u>Item No.</u>	<u>Description</u>
75	<u>MP, NMTP</u> - Total number of ARQ coefficients for panels only.
76	<u>MB, NMTB</u> - Total number of ARQ coefficients for bodies.
77	<u>NR</u> - Panel number and/or body number.
78	<u>NQ</u> - Mode number.
79	<u>NA, NARQ</u> - Number of terms (ARQ) in mode NQ, panel and/or body NR.
80	<u>N8</u> - Flag that sets origin of radial variable . If "0", the x-axis is used. If "1", inboard edge or axis of body is used as origin (see equation for τ).

The four items NR, NQ, NA, N8 are repeated for each panel or body until all panels and bodies are mentioned. Even if a panel or body is not in motion (NA = 0) the set of four items must be given.

The direction of motion of a body is given by the NZ or NY flags. Thus if body pitching is given and NY is flagged, then the body will be given a yaw. If a body can move in both the Z and Y directions (a nacelle) then it must be replaced with two bodies: one with NZ = 1 and one with NY = 1.

81	<u>N, LARQ(1)</u> - Power of x/s in the mode polynomial.
82	<u>M, LARQ(2)</u> - Power of τ/s in the mode polynomial.
83	<u>ARQ</u> - Coefficient of $(x/s)^N (\tau/s)^M$ in the mode polynomial.

Omit modal data cards if modes are not desired (Items 68 through 83).

If modes are input using the options N6=1, or N6=2, omit Items 73 through 83, and add the following data items:

<u>Item No.</u>		
84	<u>NBOX</u>	- Total number of boxes on all panels.
85	<u>B</u>	- Generalized force integration matrix (box area) X (displacement 1/4-chord of box).
86	<u>H</u>	- Displacement of 3/4-chord point normalized by semispan, h/s.
87	<u>DH1</u>	- Derivative of normalized displacement by (x/s) at 3/4-chord point of box. $d(h/s)/d(x/s)$.
88	<u>NBEL</u>	- Total number of body elements.
89	<u>B</u>	- Generalized force integration matrix (body element length) X (diameter) X (displacement 1/2-chord of element).
90	<u>H</u>	- Displacement of 1/2-chord point normalized by semispan, h/s.
91	<u>DH1</u>	- Derivative of normalized displacement by (x/s) at 1/2-chord point of element. $d(h/s)/d(x/s)$.
92	<u>DH2</u>	- $d^2(h/s)/d(x/s)^2$.

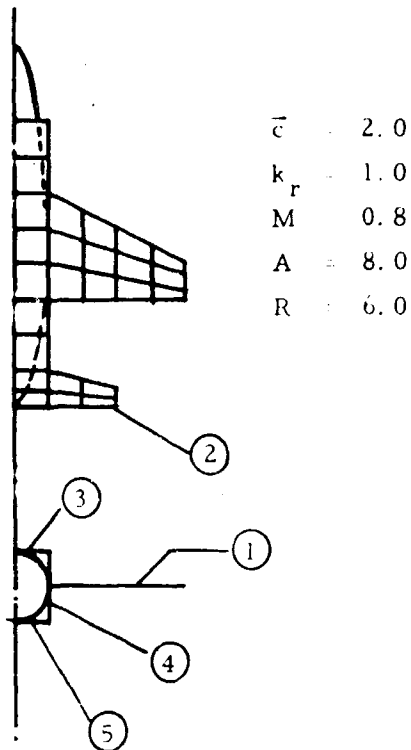
NOTE: If N6=1, Items #84 through #92 are input from cards according to the formats shown on page 8.

If N6=2, Items #84 through #92 are read from tape unit MTAPE, as one record per variable, NBOX, (B_i , $i = 1, NBOX$), (H_i , $i = 1, NBOX$) and ($DH1_i$, $i = 1, NBOX$). If bodies are also present, five more variables are read from tape unit MTAPE; again, as one record per variable: NBEL, (B_i , $i = 1, NBEL$), (H_i , $i = 1, NBEL$), ($DH1_i$, $i = 1, NBEL$) and ($DH2_i$, $i = 1, NBEL$).

For each additional mode, repeat Items #85 through #92.

2.2 Example Case

The following case is to be viewed only as an example of the proper input procedure and not as an example of an ideal or optimum idealization. The configuration is shown in Sketch 2.2-1.



Sketch 2.2-1

The fuselage is divided up into 10 equal elements for the slender body calculation. The divisions along with the radius is given as follows:

$$F_k = 0, 1.0, 2.0, 3.0, 4.0, 5.0, 6.0, 7.0, 8.0, 9.0, 10.0$$

$$RAD_k = 0, 0.7, 0.8, 0.9, 0.95, 1.0, 0.9, 0.8, 0.7, 0.5, 0.0$$

The axis of the fuselage lies on the x-axis, $Z_c = Y_c = 0$ and it is assumed that the fuselage moves only in the z-direction ($NZ = 1$). The modes of

oscillation are symmetric ($\delta = 1.0$) pitching about the middle of the fuselage and plunging

$$h/s = -5.0 + 5 x/s \text{ (where } s = 5)$$

$$h/s = 1.0$$

These modes apply to the entire configuration. The panels are numbered from 1 to 5 in Sketch 2.2-1.

The panels are given as follows:

- (1) $X_1 = 4.0, X_2 = 7.0, X_3 = 6.0, X_4 = 7.0$
 $Y_1 = 1.0, Y_2 = 5.0, Z_1 = 0, Z_2 = 0$
 $\theta = 0, 0.333, 0.666, 1.0, \tau = 0, 0.25, 0.50, 0.75, 1.0$
- (2) $X_1 = 9, X_2 = 10, X_3 = 9.5, X_4 = 10.0$
 $Y_1 = 1.0, Y_2 = 3.0, Z_1 = 0, Z_2 = 0$
 $\theta = 0, 0.5, 1.0$
 $\tau = 0, 0.5, 1.0$
- (3) $X_1 = 2.0, X_2 = 10.0, X_3 = 2.0, X_4 = 10.0$
 $Y_1 = 0, Y_2 = 1.0, Z_1 = 1.0, Z_2 = 1.0$
 $\theta = 0, 0.125, 0.25, 0.375, 0.50, 0.625, 0.75, 0.875, 0.9375, 1.0$
 $\tau = 0, 1.0$
- (4) $X_1, X_2, X_3, X_4, \theta$ same as (3)
 $Y_1 = 1.0, Y_2 = 1.0, Z_1 = 1.0, Z_2 = -1.0$
 $\tau = 0, 0.5, 1.0$
- (5) $X_1, X_2, X_3, X_4, \theta, \tau$ same as (3)
 $Y_1 = 1.0, Y_2 = 0.0, Z_1 = -1, Z_2 = -1$

The input sheets are shown on the next several pages. The output is shown on the subsequent pages.

2.2.1 Input Sheets

ENGINEER

PHONE

DATE

PUNCH NAME CARDS
73 74 75 76
H 7 W C

PROGRAM NO.	73	74	75	76	77	78	79	80	81	82	83	84	85	86	87	88	89	90	91	92	93	94	95	96	97	98	99	00	SEQ. NO.
																												7778980	
	0.0		0.5		1.0																								
	2.0		10.0		2.0																								
	1.0		1.0		2.10																								
	0.0		0.125		0.25																								
	0.75		0.875		0.9375																								
	0.0		1.0																										
	2.0		10.0		2.0																								
	1.0		1.0		3.10																								
	0.0		0.125		0.25																								
	0.75		0.875		0.9375																								

(3)

(4)

DIRECT PRINT FOR PUNCH NO UNDERPUNCHES IN SIGN FIELDS

ENGINEER _____ PHONE _____ DATE _____

PUNCH NAME CARDS 53 64 65 66 67 68 73 74 75 76 PROGRAM NO. H 7 W C

SEQ. NO.	PANEL MODES										
77	78	79	80								
11											
13											
17											
18											
19											
0.90	0.80	0.70	0.5	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
NSRPNB1	NPRINPR2	JØBNØ	JØBNØ	JØBNØ	NGUJSP	NPCNSV	NBV NY				
1.0	0.0						0.0				
LIM1	LIM2	LIM	J=1	NSTRIP							
1	3	7	9	10	12	13	14	15	16		
17	25	35	43	44	52						
NSMNT	NSQIN	NTPI	NTPM	N.PMB							
2	15	0	0	12	3						
NSRQSA	NSRQNA	NSRQNA	NSRQNA	NSRQNA	(NP)						
1	1	2	1	1	0	1	1	1	1	1	1
2	2	1	1	3	2	0	1	5	2	0	1
NSM	ARQ			i=1	MP						
0	0			5.0				0.0			5.0
1	1			5.0							5.0
0.0	0.0			5.0				0.0			7.0

2.2.2 Output

HPMC TEST CASE WING-BODY-TAIL

** ARRAY OF REDUCED FREQUENCIES **

1.000000
 REF. CHORD = 2.00000 REF. SEMI-SPAN = 5.00000 REF. AREA = 16.0
 MACH NO. = 0.80000 BETA = 0.80000

*** PANEL NO. 1 INPUT VALUES ***

X1 = 4.000000 X2 = 7.000000 Y1 = 1.000000 Z1 = 0.0
 X3 = 4.000000 X4 = 7.000000 Y2 = 3.000000 Z2 = 0.0
 NC = 4 NS = 5 NDELT = 1 NO. OF PANELS = 5

4 CIRCUMFERENCE DIVISIONS FOR PANEL 1

0.0 0.33332998E 00 0.66666498E 00 0.10000000E 01

5 SPANWISE DIVISIONS FOR PANEL 1

0.0 0.25000000E 00 0.50000000E 00 0.75000000E 00 0.10000000E 01

20 *** XI ELEMENTS FOR PANEL 1***

0.40000000E 01 0.40000000E 01 0.60000000E 01 0.70000000E 01 0.45000000E 01 0.53333244E 01
 0.61000000E 01 0.70000000E 01 0.50000000E 01 0.56666594E 01 0.63333340E 01 0.70000000E 01
 0.35000000E 01 0.59999943E 01 0.75000000E 01 0.70000000E 01 0.60000000E 01 0.63333292E 01
 0.66666500E 01 0.70000000E 01

20 *** ETA ELEMENTS FOR PANEL 1***

0.10000000E 01 0.10000000E 01 0.10000000E 01 0.10000000E 01 0.70000000E 01 0.20000000E 01
 0.20000000E 01 0.20000000E 01 0.30000000E 01 0.30000000E 01 0.30000000E 01 0.30000000E 01
 0.40000000E 01 0.40000000E 01 0.40000000E 01 0.40000000E 01 0.50000000E 01 0.50000000E 01
 0.50000000E 01 0.50000000E 01

20 ZEE ELEMENTS FOR PANEL NO. 1

0.0 0.0 0.0 0.0 0.0 0.0
 0.0 0.0 0.0 0.0 0.0 0.0
 0.0 0.0 0.0 0.0 0.0 0.0
 0.0 0.0

4 *** XI-J ELEMENTS FOR PANEL 1***

0.42500000E 01 0.47500000E 01 0.52500000E 01 0.57500000E 01

C.WIGGLES FOR PANEL NO. 1 *****

0.27500000E 01 0.22499990E 01 0.17500000E 01 0.12500000E 01

*** PANEL NO. 2 INPUT VALUES ***

X1 = 9.000000 X2 = 10.000000 Y1 = 1.000000 Z1 = 0.0
 X3 = 9.500000 X4 = 10.000000 Y2 = 3.000000 Z2 = 0.0
 NC = 5 NS = 5 NDELT = 1 NO. OF PANELS = 5

3 CIRCUMFERENCE DIVISIONS FOR PANEL 2

0.0 0.50000000E 00 0.10000000E 01

3 SPANWISE DIVISIONS FOR PANEL 2

0.0 0.50000000E 00 0.10000000E 01

9 *** XI ELEMENTS FOR PANEL 2***

0.40000000E 01 0.45000000E 01 0.10000000E 02 0.92500000E 01 0.46250000E 01 0.10000000E 02
 0.95000000E 01 0.97500000E 01 0.10000000E 02

9 *** ETA ELEMENTS FOR PANEL 2***

0.10000000E 01 0.10000000E 01 0.10000000E 01 0.20000000E 01 0.20000000E 01 0.20000000E 01
 0.30000000E 01 0.30000000E 01 0.30000000E 01

9 ZEE ELEMENTS FOR PANEL NO. 2

0.0 0.0 0.0 0.0 0.0 0.0
 0.0 0.0 0.0

2 *** XI-J ELEMENTS FOR PANEL 2***

0.91250000E 01 0.97500000E 01

C.WIGGLES FOR PANEL NO. 2 *****

0.87500000E 00 0.62500000E 00

*** PANEL NO. 3 INPUT VALUES ***

X1 = 2.000000 X2 = 10.000000 Y1 = 0.0 Z1 = 1.000000

Best Available Copy

K1 = 2.000000 K2 = 10.000000 Y1 = 1.000000 Z1 = 1.000000
M1 = 10 M2 = 2 N1 = 1 N2 = 1 N3 = 5

10 ELEMENTS DIVISIONS FOR PANEL 3

0.0 0.12500000 00 0.75000000 00 0.37500000 00 0.50000000 00 0.62500000 00
0.75000000 00 0.87500000 00 0.93750000 00 0.10000000 01

2 SPANWISE DIVISIONS FOR PANEL 3

0.0 0.10000000 01

20 *** XI ELEMENTS FOR PANEL 3***

0.20000000 01 0.30000000 01 0.40000000 01 0.50000000 01 0.60000000 01 0.70000000 01
0.80000000 01 0.90000000 01 0.95000000 01 0.10000000 02 0.20000000 01 0.30000000 01
0.40000000 01 0.50000000 01 0.60000000 01 0.70000000 01 0.80000000 01 0.90000000 01
0.95000000 01 0.10000000 02

20 *** ETA ELEMENTS FOR PANEL 3***

0.0 0.0 0.0 0.0 0.0 0.0
0.0 0.0 0.0 0.0 0.0 0.0
0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01
0.10000000 01 0.10000000 01

20 ZFF ELEMENTS FOR PANEL NO. 3

0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01
0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01
0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01
0.10000000 01 0.10000000 01

1 *** XI-J ELEMENTS FOR PANEL 3***

0.20000000 01

C.WIGGLES FOR PANEL NO. 3 *****

0.80000000 01

*** PANEL NO. 4 INPUT VALUES ***

K1 = 2.000000 K2 = 10.000000 Y1 = 1.000000 Z1 = 1.000000
K3 = 2.000000 K4 = 10.000000 Y2 = 1.000000 Z2 = -1.000000
M1 = 10 M2 = 3 N1 = 1 N2 = 1 N3 = 5

10 ELEMENTS DIVISIONS FOR PANEL 4

0.0 0.12500000 00 0.25000000 00 0.37500000 00 0.50000000 00 0.62500000 00
0.75000000 00 0.87500000 00 0.93750000 00 0.10000000 01

1 SPANWISE DIVISIONS FOR PANEL 4

0.0 0.50000000 00 0.10000000 01

10 *** XI ELEMENTS FOR PANEL 4***

0.20000000 01 0.30000000 01 0.40000000 01 0.50000000 01 0.60000000 01 0.70000000 01
0.80000000 01 0.90000000 01 0.95000000 01 0.10000000 02 0.20000000 01 0.30000000 01
0.40000000 01 0.50000000 01 0.60000000 01 0.70000000 01 0.80000000 01 0.90000000 01
0.95000000 01 0.10000000 02
0.60000000 01 0.70000000 01 0.80000000 01 0.90000000 01 0.95000000 01 0.10000000 02

10 *** ETA ELEMENTS FOR PANEL 4***

0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01
0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01
0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01
0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01
0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01

10 ZLE ELEMENTS FOR PANEL NO. 4

0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01
0.10000000 01 0.10000000 01 0.10000000 01 0.10000000 01 0.0 0.0
0.0 0.0 0.0 0.0 0.0 0.0
0.0 0.0 -0.10000000 01 -0.10000000 01 -0.10000000 01 -0.10000000 01
-0.10000000 01 -0.10000000 01 -0.10000000 01 -0.10000000 01 -0.10000000 01 -0.10000000 01

2 *** XI-J ELEMENTS FOR PANEL 4***

0.20000000 01

C.WIGGLES FOR PANEL NO. 4 *****

0.80000000 01

*** PANEL NO. 5 INPUT VALUES ***

Best Available Copy

41 = 2.000000 42 = 10.000000 43 = 1.000000 44 = -1.000000
 45 = 2.000000 46 = 10.000000 47 = 0.0 48 = -1.000000
 49 = 10 50 = 2 51 = 1 52 = 1 53 = 1 54 = 1 55 = 1

10 CIRCUMFERENTIAL DIVISIONS FOR PANEL 5

0.12500000 00	0.25000000 00	0.37500000 00	0.50000000 00	0.62500000 00	0.75000000 00
0.87500000 00	1.00000000 00	1.12500000 00	1.25000000 00	1.37500000 00	1.50000000 00

20 SPANWISE DIVISIONS FOR PANEL 5

0.10000000 01	0.20000000 01	0.30000000 01	0.40000000 01	0.50000000 01	0.60000000 01	0.70000000 01	0.80000000 01	0.90000000 01
0.10000000 01	0.20000000 01	0.30000000 01	0.40000000 01	0.50000000 01	0.60000000 01	0.70000000 01	0.80000000 01	0.90000000 01
0.10000000 01	0.20000000 01	0.30000000 01	0.40000000 01	0.50000000 01	0.60000000 01	0.70000000 01	0.80000000 01	0.90000000 01
0.10000000 01	0.20000000 01	0.30000000 01	0.40000000 01	0.50000000 01	0.60000000 01	0.70000000 01	0.80000000 01	0.90000000 01

20 *** ETA ELEMENTS FOR PANEL 5 ***

0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01
0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01
0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0

20 ZEE ELEMENTS FOR PANEL NO. 5

-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01
-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01
-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01
-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01

1 *** K-I-J ELEMENTS FOR PANEL 5 ***

0.20000000 01

C-WIGGLES FOR PANEL NO. 5 *****

0.40000000 01

12 15 25 41 52

5 DIHEDRAL ANGLES FOR ALL PANELS

0.0	0.0	0.0	-0.89999994 02	0.18000000 03	
0.49374905 01	0.58541672 01	0.67708321 01	0.76841924 01	0.86024967 01	0.95212491 01
0.56474924 01	0.62708102 01	0.68541660 01	0.74218750 01	0.80024943 01	0.85841641 01
0.64511250 01	0.69890625 01	0.75091750 01	0.80218750 01	0.85416641 01	0.90624943 01
0.72500000 01	0.77500000 01	0.82500000 01	0.87500000 01	0.92500000 01	0.97500000 01
0.98750000 01	0.27500000 01	0.17500000 01	0.47500000 01	0.57500000 01	0.67500000 01
0.77500000 01	0.87500000 01	0.97500000 01	0.98750000 01	0.98750000 01	0.98750000 01
0.47500000 01	0.57500000 01	0.67500000 01	0.77500000 01	0.87500000 01	0.97500000 01
0.98750000 01	0.27500000 01	0.17500000 01	0.47500000 01	0.57500000 01	0.67500000 01
0.77500000 01	0.87500000 01	0.97500000 01	0.98750000 01	0.98750000 01	0.98750000 01

52 *** K-V ELEMENTS ALL PANELS ***

0.44791641 01	0.53458754 01	0.61175000 01	0.69374971 01	0.77024971 01	0.84749311 01	0.92416641 01
0.51258111 01	0.59791613 01	0.65475000 01	0.71541651 01	0.77024971 01	0.82416641 01	0.87500000 01
0.62416641 01	0.67161750 01	0.71541651 01	0.75416641 01	0.79024971 01	0.82416641 01	0.85416641 01
0.82500000 01	0.87500000 01	0.92500000 01	0.97500000 01	0.98750000 01	0.98750000 01	0.98750000 01
0.42500000 01	0.52500000 01	0.62500000 01	0.72500000 01	0.82500000 01	0.92500000 01	0.97500000 01
0.92500000 01	0.22500000 01	0.12500000 01	0.42500000 01	0.52500000 01	0.62500000 01	0.72500000 01
0.72500000 01	0.82500000 01	0.92500000 01	0.97500000 01	0.98750000 01	0.98750000 01	0.98750000 01
0.42500000 01	0.52500000 01	0.62500000 01	0.72500000 01	0.82500000 01	0.92500000 01	0.97500000 01
0.92500000 01	0.22500000 01	0.12500000 01	0.42500000 01	0.52500000 01	0.62500000 01	0.72500000 01
0.72500000 01	0.82500000 01	0.92500000 01	0.97500000 01	0.98750000 01	0.98750000 01	0.98750000 01

52 *** K-L ELEMENTS ALL PANELS ***

0.42499971 01	0.52499914 01	0.62499900 01	0.72499811 01	0.82499744 01	0.92499644 01	0.97499500 01
0.51444441 01	0.59332716 01	0.65000000 01	0.70499811 01	0.75499811 01	0.80499811 01	0.85499811 01
0.81250000 01	0.86250000 01	0.91250000 01	0.96250000 01	0.97500000 01	0.97500000 01	0.97500000 01
0.42500000 01	0.52500000 01	0.62500000 01	0.72500000 01	0.82500000 01	0.92500000 01	0.97500000 01
0.92500000 01	0.22500000 01	0.12500000 01	0.42500000 01	0.52500000 01	0.62500000 01	0.72500000 01
0.72500000 01	0.82500000 01	0.92500000 01	0.97500000 01	0.98750000 01	0.98750000 01	0.98750000 01
0.42500000 01	0.52500000 01	0.62500000 01	0.72500000 01	0.82500000 01	0.92500000 01	0.97500000 01
0.92500000 01	0.22500000 01	0.12500000 01	0.42500000 01	0.52500000 01	0.62500000 01	0.72500000 01
0.72500000 01	0.82500000 01	0.92500000 01	0.97500000 01	0.98750000 01	0.98750000 01	0.98750000 01

52 *** K-J ELEMENTS ALL PANELS ***

0.0 0.0 0.0 0.0 0.0 0.0
 0.1000000E 01 0.3000000E 02 -0.3000000E 00 -0.1000000E 01

DELTA-S FOR THE 10 STRIPS

0.0 0.0 0.0 0.0 0.0 0.0
 0.0 0.1000000E 01 0.1000000E 01 0.0
 0.1250000E 00 0.1750000E 00 0.0250000E 00 0.0750000E 00 0.2500000E 00 0.7500000E 00
 0.5000000E 00 0.2500000E 00 0.7500000E 00 0.5000000E 00

*** BODY NO. 1 INPUT VALUES ***

CENTER OF BODY COORDINATES X = 0.0 Z = 0.0
 YFLAG = 0 ZFLAG = 1 SHAPE COEFFICIENT = 1.000000
 BODY DIA LIMITS = 17 92

11 BODY ELEMENTS FOR BODY NO. 1

0.0 0.1000000E 01 0.2000000E 01 0.3000000E 01 0.4000000E 01 0.5000000E 01
 0.6000000E 01 0.7000000E 01 0.8000000E 01 0.9000000E 01 1.0000000E 02

11 BODY TABLE FOR BODY NO. 1

0.0 0.6999999E 02 0.7999999E 00 0.8999999E 00 0.9999999E 00 0.1000000E 01
 0.8999999E 00 0.7999999E 00 0.6999999E 00 0.5000000E 00 0.0

10 X ELEMENTS FOR ALL BODIES

0.5000000E 00 0.1500000E 01 0.2500000E 01 0.3500000E 01 0.4500000E 01 0.5500000E 01
 0.6500000E 01 0.7500000E 01 0.8500000E 01 0.9500000E 01

10 Y ELEMENTS FOR ALL BODIES

0.5000000E 00 0.1500000E 01 0.2500000E 01 0.3500000E 01 0.4500000E 01 0.5500000E 01
 0.6500000E 01 0.7500000E 01 0.8500000E 01 0.9500000E 01

10 Z ELEMENTS FOR ALL BODIES

0.0 0.0 0.0 0.0 0.0 0.0
 0.0 0.0 0.0 0.0 0.0 0.0

10 W ELEMENTS FOR ALL BODIES

0.0 0.0 0.0 0.0 0.0 0.0
 0.0 0.0 0.0 0.0 0.0 0.0

10 V ELEMENTS FOR ALL BODIES

0.1000000E 01 0.1000000E 01 0.1000000E 01 0.1000000E 01 0.1000000E 01 0.1000000E 01
 0.1000000E 01 0.1000000E 01 0.1000000E 01 0.1000000E 01

10 U ELEMENTS FOR ALL BODIES

0.1699999E 00 0.7699999E 02 0.8699999E 00 0.9299999E 00 0.9799999E 00 0.9999999E 00
 0.9699999E 00 0.7699999E 02 0.5999999E 00 0.2500000E 00

END OF PART 1 - BASIC DATA CALCULATIONS

Best Available Copy

0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0

BUOY	NUM	N	M	REQ
1	1	0	0	-5.0000
1	1	1	0	-5.0000
1	2	0	0	1.0000

THE 10 B-MATRIX ELEMENTS FOR MODE NO. 1

-0.07999999E-01	-0.10499999E-00	-0.04999999E-01	-0.55499999E-01	-0.19500000E-01	0.18999999E-01
0.50999999E-01	0.74999999E-01	0.03299999E-01	0.06999999E-01		

THE 10 WJ MATRIX ELEMENTS FOR MODE NO. 1

0.50000000E-01	-0.22500000E-02	0.50700000E-01	-0.12500000E-02	0.50000000E-01	-0.12500000E-02
0.50000000E-01	-0.25000000E-01	0.50000000E-01	-0.25000000E-01	0.50000000E-01	0.25000000E-01
0.50000000E-01	0.74999999E-01	0.50000000E-01	0.12500000E-02	0.50000000E-01	0.12500000E-02
0.50000000E-01	0.22500000E-02				

THE 10 WJ-PRIME ELEMENTS FOR MODE NO. 1

0.0	0.50000000E-01	0.0	0.50000000E-01	0.0	0.50000000E-01
0.0	0.50000000E-01	0.0	0.50000000E-01	0.0	0.50000000E-01
0.0	0.50000000E-01	0.0	0.50000000E-01	0.0	0.50000000E-01
0.0	0.50000000E-01				

COL. NO. 1 UP DEL-CP MATRIX

0.00711171E-02	-0.07900000E-02	0.00370000E-02	0.12500000E-02	0.50000000E-02	0.12449999E-02
0.23000000E-02	0.26700000E-02	0.02200000E-02	0.22000000E-02	-0.19000000E-02	0.24770000E-02
-0.23100000E-02	0.21200000E-02	-0.12500000E-02	0.15700000E-02	-0.19200000E-02	-0.31615700E-01
-0.13170000E-02	-0.02811700E-02				

COLUMN NO. 1 OF THE 1004 MATRIX

0.25000000E-01	-0.50000000E-01	0.76000000E-01	-0.11510000E-01	0.37000000E-01	-0.00412010E-00
-0.70000000E-01	-0.26700000E-01	-0.11500000E-01	-0.23000000E-01	0.04120000E-00	-0.19000000E-01
-0.13000000E-01	-0.04000000E-01	-0.11000000E-01	-0.03000000E-01	-0.04000000E-01	-0.10000000E-01
-0.00000000E-00	0.13000000E-01	-0.12000000E-01	0.21000000E-01	-0.10000000E-01	0.02000000E-01
0.10000000E-01	0.13200000E-01	0.10700000E-01	0.23000000E-01	0.15000000E-01	0.00000000E-00
0.12000000E-01	0.50000000E-01	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0

THE 10 B-MATRIX ELEMENTS FOR MODE NO. 2

0.11000000E-01	0.20000000E-01	0.11000000E-01	0.30000000E-01	0.10000000E-01	0.17000000E-01
0.31000000E-01	0.20000000E-01	0.21000000E-01	0.00000000E-02		

THE 10 WJ MATRIX ELEMENTS FOR MODE NO. 2

0.0	0.50000000E-01	0.0	0.50000000E-01	0.0	0.50000000E-01
0.0	0.50000000E-01	0.0	0.50000000E-01	0.0	0.50000000E-01
0.0	0.50000000E-01	0.0	0.50000000E-01	0.0	0.50000000E-01
0.0	0.50000000E-01				

THE 10 WJ-PRIME ELEMENTS FOR MODE NO. 2

0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0				

COL. NO. 2 OF DEL-CP MATRIX

Best Available Copy

13	0.93125	0.25000	1.00000	0.50000	0.91834800	01	0.10957275	02
14	0.85625	2.25000	1.00000	0.50000	0.18466775	01	0.20130270	01
15	0.78125	4.25000	1.00000	0.50000	0.29603653	01	-0.27007497	-01
16	0.70625	6.25000	1.00000	0.50000	0.36076598	02	0.29843506	02
17	0.63125	8.25000	1.00000	0.50000	0.40657197	01	0.30407100	02
18	0.55625	10.25000	1.00000	-0.50000	-0.52871690	-01	-0.13068891	-01
19	0.48125	12.25000	1.00000	-0.50000	-0.14151204	00	0.28371018	00
20	0.40625	14.25000	1.00000	-0.50000	-0.40712671	01	0.39514776	01
21	0.33125	16.25000	1.00000	-0.50000	-0.13770022	02	-0.06473562	01
22	0.25625	18.25000	1.00000	-0.50000	-0.93837099	01	-0.19497306	02
23	0.18125	20.25000	1.00000	-0.50000	-0.10407286	01	-0.29131889	01
24	0.10625	22.25000	1.00000	-0.50000	-0.24607671	01	0.27047161	-01
25	0.03125	24.25000	1.00000	-0.50000	-0.36078690	02	-0.24843613	02
26	0.93125	0.25000	1.00000	-0.50000	-0.90652194	01	-0.34040809	02
27	0.85625	2.25000	0.50000	-1.00000	-0.13315166	00	0.57133595	00
28	0.78125	4.25000	0.50000	-1.00000	-0.51851849	00	-0.09999573	00
29	0.70625	6.25000	0.50000	-1.00000	0.34275133	00	-0.13137350	01
30	0.63125	8.25000	0.50000	-1.00000	0.52644482	01	0.30017004	01
31	0.55625	10.25000	0.50000	-1.00000	-0.25449181	01	0.45314584	01
32	0.48125	12.25000	0.50000	-1.00000	-0.34635274	01	0.66281500	01
33	0.40625	14.25000	0.50000	-1.00000	-0.14527224	01	-0.46493271	-01
34	0.33125	16.25000	0.50000	-1.00000	0.42646903	00	0.14574160	02
35	0.25625	18.25000	0.50000	-1.00000	-0.95462464	01	0.11491247	02

STEP NO.	NODE NO. 1			LEFT COEFFICIENT	RIGHT COEFFICIENT
	X	Z	YJS		
1	1.5000	0.0	0.5000	11.070717	29.141011
2	2.5000	0.0	0.5000	10.319186	22.614761
3	3.5000	0.0	0.5000	22.449479	20.744045
4	4.5000	0.0	0.4000	11.999816	23.635274
5	1.5000	0.0	0.5000	62.114027	172.170753
6	2.5000	0.0	0.5000	87.183362	126.249636
7	0.5000	1.0000	0.1000	1.526412	-3.719916
8	1.0000	0.5000	0.2000	7.248824	7.191329
9	1.0000	-0.5000	0.2000	-7.248796	-7.191346
10	0.5000	-1.0000	0.1000	-1.524467	3.719467

BODY NO.	NODE NO. 1			LEFT COEFFICIENT	RIGHT COEFFICIENT
	X	Z	YJS		
1	0.0	0.0	0.0	0.501440	1.761999

CF = 55.901274 52.570111 CF = 0.000029 -0.000018
 CM = -125.104490 -193.006531 CM = 0.0 0.0
 CL (ROLL) = 0.0 0.3

PRESSURE NODE	INJECTION NODE	GENERALIZED FORCES		
1	1	0.66270924	01 0.39491691	02
1	2	7.14688185	02 0.16496627	02

Best Available Copy

MODE NO. 1			LIFT COEFFICIENT		MOMENT COEFFICIENT		
STRIP NO.	Y	Z	Y35				
1	1.5000	0.0	0.0000	-12.532215	25.175320	8.659061	-10.182670
2	2.5000	0.0	0.0000	-8.600985	26.751851	5.662626	-11.602909
3	3.5000	0.0	0.0000	1.515256	26.006271	2.821109	-10.609795
4	4.5000	0.0	0.0000	2.626390	18.026368	1.525780	-6.900922
5	5.5000	0.0	0.0000	16.754325	52.986598	-1.789643	-14.150756
6	6.5000	0.0	0.0000	10.167522	51.065338	-5.610915	-13.999779
7	7.5000	1.0000	0.0000	2.021996	0.018154	-1.375269	-0.026756
8	8.5000	0.5000	0.0000	-0.155633	4.963908	-0.261762	-2.970565
9	9.5000	-0.5000	0.0000	0.155633	-4.963892	0.261755	2.970549
10	0.5000	-1.0000	0.0000	-2.021951	-0.018154	1.375241	0.026761

MODE NO. 2			LIFT COEFFICIENT		MOMENT COEFFICIENT		
STRIP NO.	Y	Z	Y35				
1	0.0	0.0	0.0	-1.929580	-0.000000	0.004467	0.189516
CL 1	-5.264211	55.211779	CL 2	0.130049	0.000325		
CM 1	10.027156	-111.647174	CM 2	0.0	0.0		
CF 1	0.0	0.0					

PRESSURE GRID	COLLECTOR GRID	GENERALIZED FORCES	
2	1	3.67429809E 01	0.15979133E 02
2	2	-0.26394281E 01	0.10773367E 02

Best Available Copy

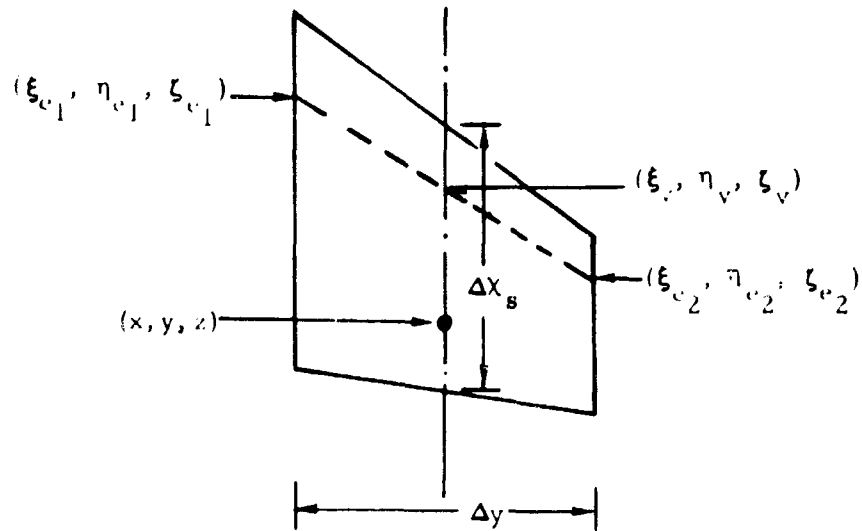
2.3 Blank and Labeled Common Blocks

2.3.1 Blank Common Block

Item No.	Mnemonics	Dimension	Symbol	Source	Description/Equation
1	NCNSM1, NBØX*			PART 1	Total number of boxes (points) on all lifting surfaces (panels) $\sum_{i=1}^N (nc_i - 1) (ns_i - 1),$ where N = NØPAN = number of panels
2	NB			MAIN	Total number of bodies - input
3	NDELT		δ	MAIN	Symmetry flag - input
4	NDATA, N3			MAIN	Data flag - input
5	NØPAN, NP			MAIN	Total number of panels - input
6	IQ		r	initialized in MAIN,	Index of receiving point
7	IR		s	used in PART 2	Index of sending point
8	JSPECS, JSP		ε	MAIN	Second symmetry flag - input
9	FMACH, M		M	MAIN	Mach Number - input
10	NCARAY	20		PART 1	Array of the number of the chordwise divisions for all panels

* A slash through the O indicates the letter O rather than the number zero.

Item No.	Mnemonics	Dimension	Symbol	Source	Description/Equation
11	NSARAY	20		PART 1	Array of the number of the spanwise divisions for all panels
12	NBARAY	20	nba_i	PART 1	$nba_i = \sum_{j=1}^i (nc_j - 1) (ns_j - 1),$ where $i = 1, N\emptyset PAN$
13	ACAP, REFARA		A, S	MAIN	Reference area - input
14	B2, S		b/2, s	MAIN	Reference semispan-input
15	FL, REFCHD		\bar{c} , $2b_r$	MAIN	Reference chord - input
16	BETA		β	MAIN	$\beta = \sqrt{1 - M^2}$, where M = Mach Number
17	PI		π	MAIN	$\pi = 3.1415926$, built-in constant
18	KR, RFREQ		k_r	MAIN	Reduced frequency - input as RFREQ(20)
19	KRDBR		k_r/b_r	MAIN	Normalized k_r
20	GMA	50	γ_i	PART 1	Array of dihedral angles for all panels in radians; $i = 1, N\emptyset PAN$
21	X	400	x	PART 1	Array of x-coordinates of receiving points (3/4 chord point of box) - see sketch on next page.



Sketch 2.3-1

Item No.	Mnemonics	Dimension	Symbol	Source	Description/Equation
22	Y	400	y	PART 1	Array of y-coordinates of receiving points
23	ZZ	400	z	PART 1	Array of z-coordinates of receiving points
24	Z1	400	ξ_{e1}	PART 1	Array of x-coordinates
25	P1	400	η_{e1}	PART 1	Array of y-coordinates
26	ZZ1	400	ζ_{e1}	PART 1	Array of z-coordinates
					} of inboard sending points
27	Z2	400	ξ_{e2}	PART 1	Array of x-coordinates
28	P2	400	η_{e2}	PART 1	Array of y-coordinates
29	ZZ2	400	ζ_{e2}	PART 1	Array of z-coordinates
					} of outboard sending points

Item No.	Mnemonics	Dimension	Symbol	Source	Description/Equation
30	EV	400	ξ_v	PART 1	Array of x-coordinates
31	PV	400	η_v	PART 1	Array of y-coordinates
32	ZV	400	ζ_v	PART 1	Array of z-coordinates
					} of center-line sending points
33	SDELX	400	Δx_s	PART 1	Array of average chordlengths of all boxes.
34	DELY	400	Δy	PART 1	Array of box widths.
					See sketch above for items no. 21-34.

The Blank Common is used by the following subroutines of the program: MAIN, PART1, GENQ, AUGW and PART2.

2.3.2 Labeled Common Blocks

Common Name	Source	User Subroutines
/BØDY/	PART1	GENQ, AUGW, GENF
/MØDE/	GENQ	AUGW
/DLM/	KERNEL	INCRØ
/NTPS/	MAIN	GENB, GENW, GENQ, PART2, SSLEÇX, CXSS, AIC, AERØ, FINAL, GENF
/PIGW/	PART1	GENB
/XXZ/	PART1	FINAL, GENF
/XYZ/	PART1	GENQ, AERØ
/YZY/	GENB	AERØ, FINAL

The following is a detailed description of the contents of each labeled common block.

Common Name	Mnemonics	Dimension	Symbol	Source	Description/Equation
BØDY	R0	100	R_{oj}	PART1	Body radius distribution
	R0P	100	R'_{oj}	PART1	x-derivative of body radius distribution
	NBEA	20	nbe_i	PART1	$nbe_i = \sum_{j=1}^i (nf_j - 1),$ where nf_j is the number of body endpoints for the j-th body; $i = 1, NB$

Common Name	Mnemonics	Symbol	Description
MØDE	N8(80)	N8	Flag for spanwise variables
	NARQ(80)	NA	Number of ARQ values for all panels and modes
	LARQ(120, 2)	(N, M)	Array of subscripts of all ARQ's
	ARQ(120)	ARQ	Array of ARQ's
	H(400)	[h]	[h] matrix
	DH1(400)	[dh/dx]	[dh/dx] matrix
	DH2(400)	[d ² h/dx ²]	[d ² h/dx ²] matrix
			} input for special cases only
			See Input Data, Sec 2.1.2

Common Name	Mnemonics	Symbol	Source	Description/Equation	
DLM	K10	$K_1^{(s)}$	KERNEL	Planar part of	} the steady contribution to the Kernel
	K20	$K_2^{(s)}$		Nonplanar part of	
	K1RT1	$Re \bar{K}_1$		$Re (\Delta K_1) T_1$	} Unsteady part of Kernel
	K1IT1	$Im \bar{K}_1$		$Im (\Delta K_1) T_1$	
	K2RT2P	$Re \bar{K}_2$		$Re (\Delta K_2) T_2^*$	
	K2IT2P	$Im \bar{K}_2$		$Im (\Delta K_2) T_2^*$	
	K10T1			$K_1^{(s)} T_1$	
	K20T2P			$K_2^{(s)} T_2^*$	
E2		e^2	INCRØ	e semi-width of sending box	

Common Name	Mnemonics	Dimension	Symbol	Source	Description/Equation
NTPS	NTP1			MAIN	Variable names for logical tape units used in program. For present assignment of units see Sec. 2.4 (Logical Tape Units)
	NTP2				
	NTP3				
	NTP4				
	NTP5				
	NTP6				
	NTP7				
	NTP8				
	NTP9				
	NTP10				
PIGW	CT1	20	\tilde{C}_1	PART1	Array of chord-lengths for inboard edge of all panels; $\tilde{C}_1 = X_2 - X_1$
	CT2	20	\tilde{C}_ϕ		Array of chord-lengths for outboard edge of all panels; $\tilde{C}_\phi = X_4 - X_3$
	TS	50	τ_{avg}		$\tau_{avg} = \frac{\tau_1 + \tau_{i+1}}{2}$ <p>where τ_i is the i-th fractional spanwise division of panel; $i = 1, NSTRIP$</p>

Common Name	Mnemonics	Dimension	Symbol	Source	Description/Equation
XXZ	X ϕ C	400	x/c	PART1	x-location of load points as a fraction of local chord
	XIJ	50	x _{le}		x-coordinate of leading edge for center of strips
XYZ	YS	50		PART1	y-coordinate of centerline of strip
	DELYS	50			Δy for strip
	ZS	50			z-coordinate of centerline of strip
	DELZS	50			Δz for strip
	FIAMMA	50			Dihedral angle for strip, in radians
	CWIG	50			Average chord length of strip
YXY	X1A*	50		GENB	(Alternative #1 AIC) Elastic axis (ALT. #2) Location of first control point.
	X3A*	50			Location of third control point.
	X5A*	50			Location of rotation axis for control surface
	X7A*	50			Location of rotation axis for tab
	X2A*	50			Location of second control point

* All coordinates given in fractions of local chord

2.4 LOGICAL TAPE UNITS

Tape Number		User Subroutines
Symbol	Present Assignment	
NTP1	1	MAIN, GENQ, PART2, SSLECX, SØLVIT, QUAS, FUTSØL
NTP2	13	MAIN, QUAS, FUTSØL
NTP3	3	MAIN, GENW, GENQ, SSLECX, CXSS, SØLVIT, QUAS, FUTSØL, AIC, FINAL, GENF
NTP4	4	MAIN, GENQ, CXSS, SØLVIT, QUAS, FUTSØL
NTP7	12	AUGW, GENF
NTP8	8	MAIN, GENQ, QUAS, FUTSØL
NTP9	9	MAIN, GENQ, SØLVIT, QUAS, FUTSØL, AIC, AERØ
NTP10	10	MAIN, GENB, GENW, GENQ, SSLECX, CXSS, AIC, FINAL, GENF
LTAPE	11	QUAS, FUTSØL
MTAPE	7	GENQ
NAT	NTP2	QUAS, FUTSØL
RHSTAP	4	QUAS, FUTSØL

Logical tape units 5 and 6 are used throughout the program as the standard READ and WRITE units respectively.

Logical tape units MTAPE, NTP9 and LTAPE can be specified as master tapes. MTAPE can be used as modal input tape for special cases; see 2.5.6.4 for pertinent information. NTP9 is used for saving the AIC supercolumns when this is desired - see 2.5.19.6. LTAPE contains the quasi-inverse matrix for future solutions of cases with new modes only - see 2.5.17.6.

Best Available Copy

2.5 DESCRIPTION OF SUBROUTINES

Program H7WC is operational on the IBM 7094 and 360/65, and the GE 635 computers. The main part of the program controls all the twenty-one subroutines of the program: PART1, GENB, GENW, GENQ, AUGW, TKER, PART2, INCRØ, KERNEL, IDF1, IDF2, SNPDF, SSLECX, CXSS, SØLVIT, QUAS, FUTSØL, AIC, AERØ, FINAL and GENF. The following is a description of each of the subroutines in the order of their use within the program.

2.5.1 MAIN

Functional Description

The MAIN part of the program reads the header card for each data set, the Mach number, the reference area, the reference semi-span and the reference chord as well as the control flags that define the various types of cases. MAIN also reads the array of reduced frequencies, and in case of an AIC-type analysis, additional control data is read, as well as the gust data, if computation of the harmonic gust coefficients is also desired. In case of a modal-type analysis, the control data is read for the modes to be used. Depending on the type of the case, MAIN selects the appropriate subroutines in an overall frequency-loop, except for PART1 and GENB (latter in case of AIC's only), which are independent of the reduced frequency.

The following is a summary of all data items used in MAIN, grouped according to their source. Detailed description of each input data item is given in Sec. 2.1.2. Other data items are described in Sec. 2.3 (Contents of Common) and/or Sec. 2.4 (Logical Tape Units) as indicated below.

Data from Cards

FMACH, ACAP, FL, B2, NDEL T, NP, NB, NRF, NCØRE, N1, N2, N3, N4, RFREQ(20), NSTRIP, NB1, NB2, NPR1, NPR2, NPR3, JØBNØ, JØBNØ2, NGUST, JSPECS, NPC, NSV, NBV, NYAW, GZRØ, XZRØ, VEL, WG, LIM(50, 3), NMD, NTA, N5, N6, N7, NTP1, NTPM, NMTP, NMTB.

See Sec. 2.1.2 for description.

Data from Common

NCNSM1, NCARAY(20), NSARAY(20), NBARAY(20), GMA(50), X(400), Y(400), ZZ(400), Z1(400), P1(400), ZZ1(400), Z2(400), P2(400), ZZ2(400), EV(400), PV(400), ZV(400), SDELX(400), DELY(400) - from Blank Common.

YS(50), DELYS(50), ZS(50), DELZS(50), FGAMMA(50), CWIG(50) - from Common /XYZ/.

X1A(50), X3A(50), X5A(50), X7A(50), X2A(50) - from Common /YZY/.

XØC(400), CWIG(400) - from Common /XXZ/.

CT1(20), CT2(20), TS(50) - from Common /PIGW/.

R0(100), R0P(100), NBEA(20) - from Common /BØDY/.

K10, K20, K1RT1, K1IT1, K2RT2P, K2IT2P, K10 I1, K10 T2P, E2 - from Common /DLM/.

See Sec. 2.3 for details.

Data from Tapes

MNEMONICS	SYMBOL	TAPE NAME	DESCRIPTION
AUGM(530)	$[A^{ww}]$	NTP1	Matrix of complex downwash factors, read one row at a time. Dimension of matrix is $NBØX \times NBØX$.
WQ(130)	$[W_q]$	NTP3	Matrix of complex polynomial mode right-hand sides, read one row (of length NMD) at a time. Dimension of matrix is $NBØX \times NMD$.
NTPØLD		LTAPE	Identification number of case for which the quasi-inverse of the matrix of downwash factors was saved on LTAPE in a previous run. For special cases only.

See Sec. 2.4 for present assignment of logical tape units.

Data to Common

NB, NDEL T, NDATA, NØPAN, IQ, IR, JSPECS, FMACH, ACAP, B2, FL, BETA, PI, KR, KRDBR - to Blank Common.

NTP1, NTP2, NTP3, NTP4, NTP5, NTP6, NTP7, NTP8, NTP9, NTP10 - to Common/NTPS/.

See Sec. 2.3 for details.

Data to Tapes

MNEMONICS	SYMBOL	TAPE NAME	DESCRIPTION
NTP1		LTAPE	Identification number of current case, used to identify the quasi-inverse that is to be saved on LTAPE. For special cases only, when the input data N5 is set to 1.
AUGM(530)	$[A^{ww}; W_q]$	NTP4	Augmented downwash matrix: matrix of downwash factors augmented by the matrix of polynomial mode right hand sides written one row (of length NBOX + NMD) at a time. Dimension of matrix is NBOX x (NBOX + NMD).

2.5.2 SUBROUTINE PART1 (JCUM, CØEFP, CØEFB, YIN, ZIN)

Functional Description

Subroutine PART1 reads the geometry input of the idealized aircraft, the desired number of spanwise and chordwise divisions per panel, and the fractional location of the division lines. The panels are subdivided into boxes, and arrays of coordinates are generated for all boxes of all panels. In the case when body interference is also desired, PART1 reads the geometry data for all bodies and generates the necessary body-coordinate arrays. The computed arrays are made available for subsequent calculations through the Blank Common block placed in the

appropriate user subroutines. Additional items of computed geometry are transmitted to MAIN via the argument list of PART1, and to several other subroutines via the Labeled Common blocks /PIGW/, /XXZ/ and /XYZ/.

Data from Cards

XCAP(4), YCAP(2), ZCAP(2), NC, NS, CØEFP(20), TH(50), TAU(50), ZSC, YSC, NF, NZ, NY, CØEFB(20), F(50), RAD(50).

See Sec. 2. 1. 2 for details.

Data from Common

NB, NDELTA, NDATA, NØPAN, IQ, IR, JSPECS, FMACH, ACAP, B2, FL, BETA, PI, KR, KRDBR - from Blank Common Block.

See Sec. 2. 3 for details.

Data from Tapes

None

Data to Common

NCNSM1, NCARAY(20), NSARAY(20), NBARAY(20), GMA(50), X(400), Y(400), ZZ(400), Z1(400), P1(400), ZZ1(400), Z2(400), P2(400), ZZ2(400), EV(400), PV(400), ZV(400), SDELX(400), DELY(400) - to Blank Common block.

YS(50), DELYS(50), ZS(50), DELZS(50), FGAMMA(50), CWIG(50) - to Common /XYZ/.

XØC(400), XIJ(50) - to Common /XXZ/.

CT1(20), CT2(20), TS(50) - to Common /PIGW/.

R0(100), R0P(100), NBEA(20), BGMA(20) - to Common /BØDY/.

Data to Tapes

None

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION/EQUATION
JCUM, NSTRIP		Number of strips on all panels
CØEFP(20)		Array of panel-coefficients - input
CØEFB(20)		Array of body-coefficients - input
YIN(20)		Array of y-coordinates of inboard edge of panels
ZIN(20)		Array of z-coordinates of inboard edge of panels

2.5.3 SUBROUTINE GENB (EV, SDELX, DELY, X, NSTRIP, BETA, IMX, NPC, NP, NSARAY, LIM)

Functional Description

Subroutine GENB reads the AIC-type mode flags for all panels of the lifting surfaces. It also reads the array of fractional chord lengths for the various control points and hinge line locations at the inboard edge and at the outboard edge of each panel. Subroutine GENB computes arrays of coordinates to be used in GENB and GENW and saves them on logical tape unit NTP10. Depending on the setting of the flag NPC (input data), subroutine GENB computes the integration matrix [B] according to Alternative 1 (NPC = 0) or Alternative 2 (NPC = 1). (See Eq. 2.5-12, Pt I, Vol I.) The [B] matrix is saved on logical tape unit NTP10 in a partitioned format, and is used in subroutine AIC for the computation of the AIC supercolumns* and harmonic gust coefficient columns, when latter is also desired.

Data from Cards

NØP(20), IS(20, 7), XHI (20, 7), XHØ (20, 7).

* A supercolumn of a matrix is a matrix-column whose elements are the columns of the original matrix taken in order; i. e., the second column is below the first, the third column is below the second, etc. A square matrix of order N becomes a supercolumn matrix of size $N^2 \times 1$.

See Sec. 2.1.2 for details.

Data from Common

NTP1, NTP2, NTP3, NTP4, NTP5, NTP6, NTP7, NTP8, NTP9,
NTP10 - from Common /NTPS/.

CT1(20), CT2(20), TS(50) - from Common /PIGW/.

Data from Tapes

None

Data to Common

X1A(50), X3A(50), X5A(50), X7A(50), X2A(50) - to Common /YZY/.

Data to Tapes

MNEMONICS	SYMBOL	TAPE NAME	DESCRIPTION
NSTRIP			Total number of strips on all lifting surfaces (panels)- input data.
NSIZ B(j, 1)			Number of AIC-modes for strip j
NSIZ B(j, 2)			Number of boxes in strip j
NSIZ B(50, 2)			
LIM(50, 3)		NTP10	Aerodynamic box limits - input
XISLCT(50, 10)			AIC-type modal flags - input
XH(50, 7)			Array of AIC-type reference coordinates
X(400)	x		} See Blank Common
SDELX(400)	Δx		
B(30, 10)	[B]		Integration matrix for the Aerodynamic Influence Coefficients (AIC's).

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
EV(400)	η_v	} See Blank Common
SDELX(400)	Δx_s	
DELY(400)	Δy	
X(400)	x	
NSTRIP		Total number of strips - input
BETA	β	See Blank Common
IMX		Maximum number of rigid body modes for case
NPC		Camber flag - input
NP, NØPAN		Number of panels - input
NSARAY(20)		Array of the number of spanwise divisions (NS) for all panels
LIM(50, 3)		Aerodynamic box limits for all strips - input

2. 5. 4 SUBROUTINE GENW (KRDBR, NCSFTØ, NPC, NGUST, WG, GZRØ, XZRØ)

Functional Description

Subroutine GENW computes the AIC-type substantial derivative matrix, [W] described in Sec. 2. 5. 1*. Depending on the setting of the flag NPC (input data), the [W] matrix is computed according to Alternative 1 (NPC = 0) or Alternative 2 (NPC = 1). The matrix is saved for subsequent calculations on logical tape unit NTP3 in a partitioned format, described in Sec. 2. 5. 2*. In the case when computation of the harmonic gust coefficient is also desired, subroutine GENW also computes the gust downwash vector, and saves it on logical tape unit NTP3.

*See Part I, Vol I

Subroutines GENB and GENW are used only if an AIC-type analysis is desired.

Data from Cards

None

Data from Common

YS(50), DELYS(50), ZS(50), DELZS(50), FGAMMA(50), CWIG(50) – from Common /XYZ/ .

Contents of Common /NTPS/.

Data from Tapes

NSTRIP, NSIZB(50, 2), LIM(50, 3), XISLCT(50, 10), XH(50, 7), X(400), SDELX(400) – from NTP10, written in GENB.

Data to Common

None

Data to Tapes

MNEMONICS	SYMBOL	TAPE NAME	DESCRIPTION
GRHS(400)	$\begin{Bmatrix} W \\ \text{gust} \end{Bmatrix}$	NTP3	<p>Gust right hand sides (downwash vector) $W_{\text{gust}_i} = \cos(\Gamma_g - \gamma_i) \cdot \exp(-i \frac{k_r}{b_r} (x_i - x_o))$, where</p> <p>$\Gamma_g$ = gust field declination from the vertical – input</p> <p>γ_i = dihedral angle of lifting surface at point i</p> <p>k_r = reduced frequency – input</p> <p>b_r = reference semichord</p> <p>x_i = x-coordinate of downwash point i</p> <p>x_o = gust reference coordinate – input</p>

MNEMONICS	SYMBOL	TAPE NAME	DESCRIPTION
W(30,10)	$[W_{AIC}]$	NTP3	Partitioned AIC-type downwash matrix. See Sec. 2.5.2*

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
KRDBR		k_r/b_r
NCSFTØ, NBØX		Total number of boxes on all panels
NPC		Camber flag
NGUST		Gust flag
WG	W_g	Harmonic gust amplitude
GZRØ	Γ_g	Gust angle of declination
XZRØ	x_o	Gust reference coordinate

} input

*See Part I, Vol I

2.5.5 SUBROUTINE GENQ (N4, N5, N6, NRF, JRF, NBE, NMD,
NMTP, NMTB, NCØRE, KDØ2, YIN,
ZIN, CØEFP, CØEFB)

Functional Description

Subroutine GENQ generates the normalwash boundary condition for all lifting surface elements. It also generates the integration matrix used to obtain the generalized forces. The normalwash w_{ri} (r ranges over all lifting surface box receiving points while i ranges over all modes) is written on tape row by row (r = 1, i = 1, 2, ..., NMD, r = 2, i = 1, 2, ..., NMD, etc.) to facilitate its use with subroutine SØLVIT. However, if the number of modes times the number of unknowns (NCØRE) exceeds the size of the working array in SØLVIT, and/or if the input flag N5 is set to 1, the W matrix is written on tape in column order, preceded by the number of modes (NMD), ready for use by subroutines QUAS and FUTSØJ. The integration matrix B_{ri} is written on tape in column order.

The formula for w used is given by equations 2.3-2* and 2.3-3* for cases without bodies, and by equation 2.2-13* for body cases; the contribution of the body effect to the normalwash, w_{B_r} , is computed in subroutine AUGW, which is called from subroutine GENQ for body cases only. w_{B_r} is computed one column at a time, and is transmitted to GENQ via the argument list of subroutine AUGW. The total normalwash matrix for body cases, $(w_r - w_{B_r})_i$, is then written on tape in either row, or column order depending on the size (NCØRE) of the problem and/or on the setting of the input flag N5.

Data from Cards

NR, NQ, NARQ(80), N8(80), LARQ(120, 2) and ARQ(120) for standard polynomial cases (N6 = 0).

When modes are input from cards (N6 = 2), the data input from cards are:

NBØX, BQ(400), H(400) and EH1(400).

*See Part I, Vol I

Data from Common

Contents of Blank Common and the labeled common blocks
/NTPS/ and /BODY/.

Data from Tapes

$N6 = 0$ cases:

None, if $NRF = 1$. If $NRF > 1$ and $JRF > 1$ (see Sec. 2.5.5.), the following data items are read from tape NTP10:

BQ(400), N8(80), NARQ(80), LARQ(120, 2), and ARQ(120).

$N6 = 1$ cases: modes are input from tape as follows.

NBØX, BQ(400), H(400) and DH1(400) from tape MTAPE; present assignment is NTP8 = 8.

Data to Common

N8(80), NARQ(80), LARQ(120, 2), and ARQ(120),
H(400), DH1(400) and DH2(400) to common /MODE/.

Data to Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
BQ(400)	$[B_q]$	NTP10	Integration matrix for modal analysis written in column order. Dimension of matrix is $NTØT \times NMD$, where $NTØT$ is the total number of elements (e.g., $NTØT = NBØX$, if there are no body elements, and $NTØT = NBØX + NBE$ otherwise, where $NBE =$ total number of body elements).
NMD	NM	NTP4	Number of modes; written only if $N6 = 1$ and/or $NCØRE \geq 4000$.

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
WQ(400)	$[W_q]$	NTPW	<p>Matrix of polynomial mode right hand sides (normalwash). Dimension of matrix is $NB\emptyset X \times NMD$. The matrix $[W_q]$ is written on tape in row order, unless $N5 = 1$, and/or $NC\emptyset RE \geq 4000$; in latter case, $[W_q]$ is written on tape in column order.</p> <p>Note that NTPW = NTP3 in case of row order, and NTPW = NTP4 in case of column order.</p>
BQ(400)	$[B_q]$	NTPB	<p>The following items are written on tape only if more than one reduced frequency is prescribed for the case ($NRF > 1$).</p> <p>See Input Data, 2.1.2</p> <p>Note that NTPB = NTP2 for cases without bodies, and NTPB = 10 for body cases.</p>
N8(80)			
NARQ(80)	NA		
LARQ(120, 2)	(N, M)		
ARQ(120)	ARQ		

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
N4		Print flag - input
N5		Flag to save quasi-inverse - input
N6		Modal input data flag - input
NRF		Number of reduced frequencies for case - input
JRF		Sequence number of reduced frequency currently used in the frequency do-loop

MNEMONICS	SYMBOL	DESCRIPTION
NBE		Total number of body elements
NMD	NM	Number of modes - input
NMTP	MP	Total number of ARQ - values for all panels and modes - input
NMTB	MB	Total number of ARQ - values for all bodies and modes - input
NCØRE		Total number of unknowns for case - input
KDØ2	KD/2	4000 - built in constant (MAIN)
YIN(20)	Y1	Array of y-coordinates of inboard edge of panels
ZIN(20)	Z1	Array of z-coordinates of inboard edge of panels.
CØEFP(20)	} coef	Array of modal coefficients for panels
CØEFB(20)		Array of modal coefficients for bodies
		} See Eq.(2. 3-3)

2.5.6 SUBROUTINE AUGW (NMD, NBOX, NBE, NMTB, N4, N6, NRF, JRF, J, CØEFB, ASUM)

Functional Description

This subroutine gives the incremental normalwash, w_{Br} , at all lifting surface boxes, r , due to the axial singularities on all bodies, B . Slender body theory is used to obtain the strength of the axial singularities. The formulae used in this subroutine are given by Equations(2.2-8)through (2.2-10)* The incremental normalwash values, w_{Br} , are transmitted to subroutine GENQ via the argument list under the mnemonic name ASUM one column (for one mode) at a time. This subroutine also computes the interaction matrix of all body elements for the calculation of the generalized forces for body cases.

Subroutine AUGW is bypassed for data without body elements.

*See Part I, Vol I

Data from Cards

None for the special case when $N6 = 1$ (data input from tape MTAPE).

For standard polynomial cases ($N6 = 0$) data input from cards is same as for subroutine GENQ, except that data is now read for body elements instead of panel boxes.

Data from Common

Contents of Blank Common, and the labeled common blocks /BØDY/ and /MØDE/.

Data from Tapes

None for standard polynomial mode cases. For the special case when the modal data flag $N6 = 1$, the following data items are input from tape:

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
NBE			Number of body elements
BB(80)	$[B_q]$	NTP8	Integration matrix for the body elements
H(400)	$[h]$		$[h]$ matrix
DH1 (400)	$[dh/dx]$		$[dh/dx]$ matrix
DH2(400)	$[d^2h/dx^2]$		$[d^2h/dx^2]$ matrix
			The above three matrices are read in column order. Dimension of matrices is NBE x NMD.

Data to Common

None

Data to Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
BB(80)	B_q	NTP8	One column of the integration matrix for body elements written on tape for one mode; length of column is NBE.
N8(80)	N8	NTPB	The following items are written on tape only if more than one reduced frequency is prescribed for case (NRF > 1) See Input Data 2.1.2
NARQ(80)	NA		
LARQ(120, 2)	(N, M)		
ARQ(120, 2)	ARQ		

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
NMD	NM	Number of modes - input
NBØX		Total number of boxes on all lifting surfaces
NBE		Total number of body elements
NMTB	MB	Total number of ARQ values for all bodies and modes - input

MNEMONICS	SYMBOL	DESCRIPTION
N4		Print flag for writing the [AP ^Z], [AP ^Y] matrices
N6		Modal input flag
NRF		Number of reduced frequencies for case
JRF		Sequence number of current frequency in the frequency do-loop
J		Sequence number of current mode in modal do-loop
CØEFB(20)	coef	Array of scale factors for body deflection modes.
ASUM(400)		Incremental [W] due to body interference

} input

2.5.7 SUBROUTINE TKER (X0, Y0, Z0, KR, BR, GAMS, GAMSIG, KKR, KKI, FMACH)

Functional Description

Subroutine TKER evaluates the total unsteady kernel for one receiving-point/sending-point combination for body cases only. (See Eq. A. 5)* Subroutine TKER is essentially the same as subroutine KERNEL, except that the total, rather than the incremental oscillatory kernel is computed here, and that the planar and nonplanar parts of the kernel are not separated. Subroutine TKER is called from subroutine AUGW; it is bypassed in the case when the input data contains no bodies.

Subroutine TKER uses no card-, tape-, or common-data input or output; data transmission to and from TKER is done through its argument list.

*See Part I, Vol I

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
X0	x_0	$x_i - \xi_j$ $y_i - \eta_j$ $z_i - \xi_j$
Y0	y_0	
Z0	z_0	
KR	k_r	Reduced frequency
BR	b_r	Reference semi-chord
GAMS	γ_s	Dihedral angle of sending point j
GAMSIG	γ_r	Dihedral angle of receiving point i
FMACH, M	M	Mach Number
E2		e^2 , where e is semi-width of sending box
KKR		Real part of total (steady + unsteady) kernel
KKI		Imaginary part of total kernel

2.5.8 SUBROUTINE PART2 (NYAW, NBV)

Functional Description

Subroutine PART2 is used for the calculation of the matrix of downwash factors, [D]. (See Eq. 2.1-14.*) It prepares the arguments to subroutines SNPFD and INCRØ, which do the actual computation of the downwash factor elements, and calls these in a do-loop for all boxes of all lifting surfaces. The steady part of the downwash factor elements is computed in subroutine SNPFD, and the incremental oscillatory part in subroutine INCRØ; the call to INCRØ is bypassed for steady cases ($k_r = 0.0$). Subroutine PART 2 also accounts for the effect of the (two) symmetry plane(s).

Data from Cards

None

*See Part I, Vol I

Data from Common

Contents of Blank Common, Common /NTPS/, and Common /DLM/.

Data from Tapes

None

Data to Common

None

Data to Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
AWW(400) AWWI(400)	} [A ^{ww}], [D]	NTP1	Matrix of downwash factors written on tape in row order. Dimension of matrix is NBØX x NBØX. The real parts of the matrix elements are called AWW in the program, and the imaginary parts are called AWWI.

Data in Argument List

NYAW - yaw flag
NBV - number of boxes on vertical panels } input in MAIN
(fin) of T-tail case:

2.5.9 SUBROUTINE INCRØ (AX, AY, AZ, AX1, AY1, AZ1, AX2, AY2, AZ2, GAMS, GAMSIG, LHS, IR, IØ, NBXS, NCPNB, NDBLE, NBV, DELR, DELT, FL, BETA, SDELX, DELY, KR)

Functional Description

Subroutine INCRØ prepares the argument for the subroutines KERNEL, IDF1 and IDF2. It calls subroutine KERNEL to compute the

incremental oscillatory part of the kernel, K , for each receiving-sending box combination at the three points of the bound vortex segment: at the center (K_c), at the inboard point (K_i) and at the outboard point (K_o). Since even a relatively small case requires many kernel computations (e. g. , a wing with 100 boxes, in symmetry, requires $100 \times 300 \times 2 = 60,000$ kernel values), extra programming effort was made to save on computing time. Neighboring strips within a wing panel have a common kernel on the common boundary; this fact is utilized in INCRØ so that the computing time of the kernels is reduced by about 30 percent.

After the triplet of kernels, (K_c , K_i , K_o), is obtained for one receiving-sending box combination, subroutine INCRØ computes the coefficients of the parabolas for the numerical integrations done in two parts. The planar part of the integration is done in subroutine IDF1, and the nonplanar part in IDF2.

Subroutine INCRØ uses no card-, or tape input or output; transmission of data is done through its argument list and the labeled common /DLM/.

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION/EQUATION
AX	x_o	$x - \xi_v$
AY	y_o	$y - \eta_v$
AZ	z_o	$z - \zeta_v$
AX1		$x - \xi_{e1}$
AY1		$y - \eta_{e1}$
AZ1		$z - \zeta_{e1}$
AX2		$x - \xi_{e2}$
AY2		$y - \eta_{e2}$
AZ2		$z - \zeta_{e2}$

See Sec. 2.3.1
(Blank Common)

MNEMONICS	SYMBOL	DESCRIPTION/EQUATION
GAMS	γ_r	Dihedral angle of receiving point i
GAMSIG	γ_s	Dihedral angle of sending point j
LHS		Flag; 0 when effect of right wing is computed, 1 for left-wing
IR	j	Index of sending point
IØ		An index running from 1 through NCM1, where NCM1 is the number of chordwise boxes of the strip in which the sending point lies.
NBXS		The number of boxes on the wing panel in which the sending point lies + the total number of boxes of the preceding panels.
NCPNB		The number of boxes in the first strip of the panel in which the sending point lies + the total number of boxes of the preceding panels.
NDBLE, JSPECS, JSP	ϵ	Second symmetry flag - input
NBV		Total number of boxes on vertical panels of T-tail cases - input
DELR		Real part of the incremental oscillatory downwash factor
DELI		Imaginary part of the incremental oscillatory downwash factor
FL, REFCHD	\bar{c}	Reference chord - input
BETA	β	$\sqrt{1 - M^2}$
SDELX	Δx_s	Average box-chord of sending box
DELY		2c, box width of sending point
KR	k_r	Reduced frequency

2.5.10

SUBROUTINE KERNEL (X0, Y0, Z0, KR, BR, GAMS, GAMSIG,
EPS, T1, T2, M)

Functional Description

Subroutine KERNEL evaluates the modified incremental oscillatory kernel, \bar{K} , for one receiving-sending point combination at a time. The planar and nonplanar parts of the kernel are evaluated separately and are returned to the calling program, subroutine INCRØ, via the labeled common /DLM/. (Evaluates right hand side of Eq. (2.1-13).*)

Subroutine KERNEL uses no card or tape input or output; transmission of data is done through the argument list and the labeled common /DLM/.

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION/EQUATION
X0	x_0	AX for center AX1 for inboard edge AX2 for outboard edge
Y0	y_0	AY for center AY1 for inboard edge AY2 for outboard edge
Z0	z_0	AZ for center AZ1 for inboard edge AZ2 for outboard edge
KR	k_r	Reduced frequency
BR	$b_r, c/2$	Reference semichord
GAMS	γ_r	} see 2.5.9
GAMSIG	γ_s	

*See Part I, Vol I

MNEMONICS	SYMBOL	DESCRIPTION/EQUATION
EPS	ϵ	$\epsilon = 0.00001$; built in constant from INCRØ
T1	T_1	$T_1 = \cos(Y_r - Y_s)$
T2	T_2	$T_2 = [z_o \cos Y_r - y_o \sin Y_r](z_o \cos Y_s - y_o \sin Y_s) / r^2$
M, FMACH	M	Mach number

2.5.11 SUBROUTINE IDF1(EE, E2, AT2, ETA01, ZET01, ARE, AIM, BRE, BIM, CRE, CIM, RISQX, XIIJR, XIIJI)

Functional Description

Subroutine IDF1 performs the integration of the planar parts of the incremental oscillatory kernels; see Eq. B. 9.* The result of the integration is the complex number (XIIJR, XIIJI), returned to subroutine INCRØ via the argument list of IDF1.

Subroutine IDF1 uses no card-, tape- or common input or output; data transmission to and from IDF1 is done through its argument list.

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
EE	$e, \Delta y_s / 2$	Semiwidth of sending box j
E2	e^2	
AT2	$ T_2 $	See 2.5.10
ETA01	\bar{y}	$y_o \cos Y_r + z_o \sin Y_r$
ZET01	\bar{z}	$z_o \cos Y_r - y_o \sin Y_r$

*See Part I, Vol I

MNEMONICS	SYMBOL	DESCRIPTION	
ARE	$\text{Re}(A_1)$	Coefficients of the parabola for planar part of kernel-integration	
AIM	$\text{Im}(A_1)$		
BRE	$\text{Re}(B_1)$		
BIM	$\text{Im}(B_1)$		
CRE	$\text{Re}(C_1)$		
CIM	$\text{Im}(C_1)$		
RISQX	r^2	$\frac{-z}{y} + \frac{z^2}{z}$	
XIIJR		Real part of integral	} planar contribution
XIIJI		Imaginary part of integral	

2. 5. 12 SUBROUTINE IDF2 (EE, E2, AT2, ETA01, ZET01, A2R,
A2I, B2R, B2I, C2R, C2I, RISQX,
DIJR, DIJI)

Functional Description

Subroutine IDF2 performs the integration of the nonplanar parts of the incremental oscillatory kernel; see Eqs. (B. 15) and (B. 16).^{*} The result of the integration is the complex number (DIJR, DIJJ), returned to subroutine INCRØ through the argument list of IDF2.

Subroutine IDF2 uses no card , tape or common input or output; data transmission to and from IDF2 is done through its argument list.

^{*}See Part I, Vol I

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
EE	e	} See 2.5.11
EZ	e^2	
ATZ	$ T_2 $	
ETA01	\bar{y}	
ZET01	\bar{z}	} Coefficients of the parabola for nonplanar part of kernel integration
AZR	$\text{Re}(A_2)$	
AZI	$\text{Im}(A_2)$	
BZR	$\text{Re}(B_2)$	
BZI	$\text{Im}(B_2)$	
CZR	$\text{Re}(C_2)$	
CZI	$\text{Im}(C_2)$	} See 2.5.11
R1SQX		
DIJR		
DIJI		Real part of integral } nonplanar contribution
		Imaginary part of integral }

2.5.13 SUBROUTINE SNPFD (SL, CL, TL, SGS, CGS, SGR, CGR, X0, Y0, Z0, EE, DIJ, BETA, CV)

Functional Description

Subroutine SNPFD computes the steady downwash factors for one receiving-sending box combination at a time. The result, DIJ, is returned to PART2 via its argument list. (See Eq. C.35b, Part I, Vol I.)

Subroutine SNPFD uses no card, tape, or common data input or output; data transmission to and from SNPFD is done through its argument list.

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
SL		$\sin \Lambda$, where Λ is the sweep angle of the 1/4-chord line of sending box [sweep angle of bound vortex] modified by Prandtl-Glauert rule.
CL		$\cos \Lambda$
TL		$\tan \Lambda$
SGS		$\sin \gamma_s$
CGS		$\cos \gamma_s$
SGR		$\sin \gamma_r$
CGR		$\cos \gamma_r$
X0		} See 2.5.10
Y0		
Z0		
EE	e	Semi-width of sending box
DIJ		Steady contribution to downwash factor
BETA	β	$\sqrt{1 - M^2}$
CV	$c_{avg} \Delta x_s$	Δx , average chord length of sending box

2. 5. 14

SUBROUTINE SSLECX (NCNSM1, MD, NGUST)Functional Description

Subroutine SSLECX is used with AIC-type data, and only if $NC\emptyset RE < 4000$. It reads the matrix of downwash factors $[D]$ from tape NTP1 one row at a time, and augments it with the AIC-type downwash matrix $[W]_{AIC}$ read from tape NTP3. For gust cases, when $NGUST \neq 0$, subroutine SSLECX also reads the gust downwash vector from tape NTP3. The augmented matrix $[D|W]$ is saved on tape NTP4 in row order, ready to be used by subroutine SOLVIT to obtain the solutions to the system of simultaneous linear equations represented by the matrix $[D|W]$.

Data from Cards

None

Data from Common

Contents of Common /NTPS/.

Data from Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
NSTRIP			} See 2. 5. 3
NSIZ B(50, 2)		NTP10	
LIM(50, 3)			
AUGM(530)	[A]	NTP1	Matrix of downwash factors written on tape in row order in subroutine PART2. Dimension of matrix is $NB\emptyset X \times NB\emptyset X$.
GUST(400)	$\{W_{gust}\}$	NTP3	Gust downwash vector
W(30, 10)	$[W]_{AIC}$	NTP3	AIC type downwash (normalwash). } See 2. 5. 4

Data to Common

None

Data to Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
AUGM(530)	[A W]	NTP4	Augmented downwash factor matrix

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
NCNSM1	n	Total number of boxes on all lifting surfaces (panels)
MD	m	Total number of AIC-type right-hand sides $NSTRIP$ $m = \sum_{j=1}^{NSTRIP} XISLCT_{jmax}$ <p>where $NSTRIP$ = total number of strips on all lifting surfaces, and $XISLCT_{jmax}$ = the number of elastic modes for j-th strip - whenever $NGUST = 0$. When $NGUST \neq 0$</p> $m = \sum_{j=1}^{NSTRIP} XISLCT_{jmax} + 1.$
NGUST		Gust flag - input

2.5.15

SUBROUTINE CXSS (NCNSM1, MD, NGUST)

Functional Description

Subroutine CXSS is used with AIC-type data, and only if $NC\emptyset RE \geq 4000$, and/or $N5 = 1$. It reads the AIC-type downwash matrix, $[W]_{AIC}$, from tape NTP3 and forms columns of right-hand sides. When $NGUST \neq 0$, subroutine CXSS also reads the gust downwash vector from tape NTP3; this becomes the last column of right hand sides. These are saved on tape NTP4 preceded by the number of right hand sides for the case, ready to be used by subroutine QUAS and FUTS\emptyset L to obtain the solutions.

Data from Cards

None

Data from Common

Contents of Common /NTPS/.

Data from Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
NSTRIP			} See 2.5.3
NSIZB(50, 2)		NTP10	
LIM(50, 3)			
GUST(400)	$\{W_g\}$	NTP3	Gust downwash vector
W(30, 10)	$[W]_{AIC}$	NTP3	Partitioned matrix of AIC-type downwash (normalwash) } see 2.5.4

Data to Common

None

Data to Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
NMD	m	} NTP4	Number of AIC-type right hand sides (elastic modes), including the gust right hand side whenever NGUST ≠ 0
WX(400)			Matrix of AIC-type right hand sides written in column order. Dimensions of matrix is NBØX x NMD

Data in Argument List

Same as in 2. 5. 14

2. 5. 16 SUBRØUTINE SØLVIT (ND, MD, KD, NI, NM, NØ, NW, NPR1)

Functional Description

Subroutine SØLVIT is used for cases with NCØRE < 4000, and/or N5 = 0. It solves the system of simultaneous linear equations represented by the augmented matrix $[D \mid W]$, written on tape NTP4. The solutions are saved on tape NTP3 in column order; the data on the input tape, NTP4, is not preserved.

Data from Cards

None

Data from Common

None

Data from Tapes

AUGM(530), the augmented downwash factor matrix from tape NTP4 - see 2. 5. 1 or 2. 5. 14.

Data to Common

None

Data to Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
A(4000)		NTP3	Solution matrix written in column order; dimension of matrix is NDxMD, where ND = NBØX, and MD = NMD.

Note that subroutine SØLVIT uses a single complex work array, A(4000), for all tape input and output variables.

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
ND, NCNSM1, NBØX	n	Total number of boxes on all lifting surfaces (panels) = size of the square matrix $[A^{ww}] = [D]$ = matrix of downwash factors.
MD	m	Total number of right hand sides in the system of simultaneous linear equations solved in subroutine SØLVIT.
KD		Working array size (real-variable dimension); presently set to 8000
NI		NTP4; tape number assigned to tape containing all rows of the augmented matrix $[A^{ww} W]$ - see 2.5.1 or 2.5.14
NM		NTP1; tape number, used as scratch unit
NØ		NTP9; tape number, used as scratch unit
NW		NTP3; tape number assigned to tape containing all solutions of the system of simultaneous linear equations
NPR1		Print flag for solutions - input

For present assignment of tape units see 2.4.

2.5.17 SUBROUTINE QUAS (ND, MD, KD, NI, MM, NØ, NAT, NW,
LTAPE, RHSTAP, NPI)

Functional Description

Subroutine QUAS is used in conjunction with subroutine FUTSØL for cases with NCØRE \geq 4000, and/or N5 = 1. It reads the matrix of downwash factors, [D] from tape NTP1 in row order, and as many columns of right hand sides (m, say), from tape RHSTAP, as the dimension of the work area in QUAS (3700 complex words) will permit. Subroutine QUAS computes the so called quasi-inverse of the matrix [D] and saves it on tape LTAPE; it also computes the solutions for the m right hand sides and saves these on tape NTP3.

Data from Cards

None

Data from Common

None

Data from Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
A(3700)	[D], [A ^{ww}]	NTP1	Matrix of downwash factors; see 2.5.8
NMD		NTP4	Total number of modes for case
A(3700)	[W]	NTP4	Matrix of right hand sides; see 2.5.5 or 2.5.15

Data to Common

None

Data to Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
A(3700)		LTAPE	Quasi-inverse matrix of [D]. Master tape has to be specified to preserve quasi-inverse for future analysis; otherwise tape is considered a scratch unit.
A(3700)		NTP3	Solution matrix; dimension $NB\emptyset X \times n$, where n is the number of right hand sides for which solutions were obtained.

Note that Subroutine QUAS uses a single complex work array, A(3700) for all input and output variables.

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
ND, NCNSM1, NB \emptyset X	n	Total number of boxes on all lifting surfaces
MD	m	On output MD = n = the number of right hand sides for which solutions were obtained
KD		Working array size (complex variable dimension); presently set to 3700.
NI		NTP1; tape number assigned to tape containing all rows of matrix [A ^{ww}] (matrix of downwash factors)
MM		NTP8
N \emptyset		NTP9
NAT		NTP2
		} tape numbers for scratch units

MNEMONICS	SYMBOL	DESCRIPTION
NW		NTP3, tape number for the solution matrix
LTAPE		Tape number assigned to the tape containing the quasi-inverse of [D]
RHSTAP		NTP4; tape number assigned to tape containing the matrix of right hand sides

For present assignment of tape units see Sec. 2.4

2.5.18 SUBROUTINE FUTSØL (ND, MD, KD, NL, MM, NØ, NAT, NW, LTAPE, RHSTAP)

Functional Description

Subroutine FUTSØL is used for cases with NCØRE \geq 4000, and whenever new modes only are required for a previously computed case (N1 = 1). When N1 = 0, subroutine FUTSØL is called following subroutine QUAS, and uses the quasi-inverse of the matrix [D] from tape LTAPE. Additional right hand sides are read from tape RHSTAP, and the additional solutions are saved on tape NTP3 following the previously obtained solutions.

Subroutine FUTSØL is bypassed when N5 = 1 and NCØRE < 3700.

Data from Cards

None

Data from Common

None

Data from Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
A(3700)	[W]	RHSTAP	Additional columns of right hand sides
A(3700)		LTAPE	The quasi-inverse matrix generated in QUAS

Data to Common

None

Data to Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
A(3700)		NW	The additional solutions for the additional right hand sides

For present assignment of tape units see Sec. 2.4.

Data in Argument List

Same as for QUAS - see 2.5.17, except that NI is now merely a scratch unit and does not contain the input matrix [D].

2.5.19 SUBROUTINE AIC(BR,S,KR,NBF,NBL,JØBNØ,ICØD,VEL,NGUST,NPR2)

Functional Description

Subroutine AIC is used for AIC-type cases only. It reads the solutions to the AIC-type elastic modes from tape NTP3 and the partitioned integration matrix, [B], from tape NTP10. Using these arrays subroutine AIC supercolumns according to Eq. 2.5-1*, and saves them on tape NTP9.

*See Part I, Vol I

Data from Cards

None

Data from Common

Contents of the labeled common blocks /NTPS/, /XYZ/ and /ZYZ/.

Data from Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
NSTRIP	}	NTP10	See 2. 5. 3
NSIZE(50, 2)			
LIM(50, 3)			
B(10, 30)	$[B]_{AIC}$		Integration matrix
DIJMIW(400, 4)	$[D]^{-1}[W]_{AIC}$	NTP3	Solution matrix for the AIC type right hand sides
CPGUST(400)		NTP3	Solution for the gust right hand side - read only if NGUST \neq 0.

Data to Common

None

Data to Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
CH(4, 4, 50)		NTP9	AIC matrix written one supercolumn at a time; see Eq. 2. 6-2. *

*See Part I, Vol I

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
BK	$b_r, c/2$	Reference semichord
S, REFSPN	$b/2, s$	Reference semispan - input
KR	$b/2$	Reduced frequency
NBF		First bay number
NBL		Last bay number
JØBNØ		AIC-case-identifier
ICØD		1, built in constant
VEL		Aircraft velocity
NGUST		Gust flag
NPR2		Print flag for AIC's

for which
AIC's are
written on
tape

- input

2. 5. 20 SUBROUTINE AERØ (BR, B2, KR, ACAP, NRF, NSV, NYAW,
IMX, NPC, NDEL, NPR3)

Functional Description

Subroutine AERØ is used for AIC-type cases only. It reads the AIC supercolumns from tape NTP9 and computes the stability derivatives for the rigid body modes according to Eqs. (2.4-6) - (2.4-12)*. This subroutine is primarily used for the monitoring of the AIC's. It can be bypassed by setting the data flag NPR3 to 0.

Data from Cards

None

Data from Common

Contents of the labeled commons /NTPS/, /XYZ/ and /ZYZ/.

*See Part I, Vol I

Data from Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
NSTRIP	}	NTP10	See 2. 5. 3
NSIZB(50, 2)			
LIM(50, 3)			
XISLCT(50, 7)			
CH(4, 4, 50)		NTP9	See 2. 5. 19

Data to Common

None

Data to Tapes

Usually none. However, in special cases, when dynamic stability derivatives are also required (input flag NPR3=3), tape unit 2 is used for saving the total lift-and moment coefficients computed for the steady case ($KR(1) = 0.0$). Subsequently, this information is read from tape unit 2 and is used in the calculation of the dynamic stability derivatives for the unsteady case ($KR(2) > 0.0$) - see 2. 4. 2, in Part I, Vol I.

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
BR	$b_r, c/2$	} See 2. 5. 19
B2	b_2, s	
KR	k_r	
ACAP, REFARA	A, S	Reference area - input

MNEMONICS	SYMBOL	DESCRIPTION
NRF		Number of reduced frequencies for case - input
NSV		Number of strips on vertical panels (fin) of T-tail cases; 0 otherwise - input
NYAW		Yaw flag - input
IMX		Number of rigid body modes for AIC-type case-from GENB
NPC		Camber flag
NDELT	δ	Symmetry flag
NPR3		Print flag

2. 5. 21 SUBROUTINE FINAL (BR, B2, ACAP, NSV, NBV, IMX, NYAW, NPC, NDELT, FGAMMA)

Functional Description

Subroutine FINAL is used for AIC-type cases only. It reads the solutions for the AIC-type right hand sides from tape NTP3 and computes the pressures for the rigid body modes for all boxes of all lifting surfaces. These are printed along with the box number and the fractional chordwise location of the load points (1/4-chord of box). Subroutine FINAL can be bypassed by setting the data flag NPR3 to 0.

Data from Cards

None

Data from Common

Contents of the labeled commons /NTPS/, /XXZ/, and /YZY/.

Data from Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
NSTRIP		NTP10	See 2.5.3
NSIZB(50, 2)			
LIM(50, 3)			
CPM(400, 4)		NTP3	Solution matrix for the AIC-type right hand sides

Data to Common

None

Data to Tapes

None

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
BR	b_r	See 2.5.20
B2	$b/2, s$	
ACAP, REFARA	A, S	
NSV		Total number of boxes on vertical panels (fin) of T-tail cases - input
NBV		
IMX		See 2.1.2
NYAW		
NPC		
NDELT	δ	Array of dihedral angles for all strips of all lifting surfaces.
FGAMMA(50)		

2. 5. 22

SUBROUTINE GENF (NBØX, NTØT, NMD, NDELT, NST,
LIM, FL, XCAP, XII, EV, SDELX,
BZ, Y, ZZ, XØC, NBV)

Functional Description

Subroutine GENF is used for modal type cases only. It reads the solutions for all modes from tape NTP3 and writes the pressures for all modes and all boxes along with the x, y, z coordinates of the boxes. It also computes the sectional lift and moment coefficients for all strips of all lifting surfaces as well as the total lift- and moment coefficients - see Eqs. (2.4-6)-(2.4-12)*Subroutine GENF also computes the generalized forces for all pressure modes and deflection modes; see Eq. 2.4-5*

Data from Cards

None

Data from Common

Contents of the labeled common blocks /NTPS/, /XYZ/ and /BØDY/.

Data from Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
P(400)	$\left\{ \Delta C_p \right\}$	NTP3	Solution matrix for modal type data read in column order. Dimension of matrix is NBØXxNMD.
P(400)	$\left\{ \Delta C_p^{(f)} \right\}$	NTP7	Matrix of pressures for all body elements and modes read in column order. Dimension of matrix is NBExNMD. This data is read for body cases only.
B(400)		NTP10	Integration matrix for modal type data read in column order. Dimension of matrix is NTØTxNMD where NTØT = NBØX + NBE.

Data to Common

None

*See Part I, Vol I

Data to Tapes

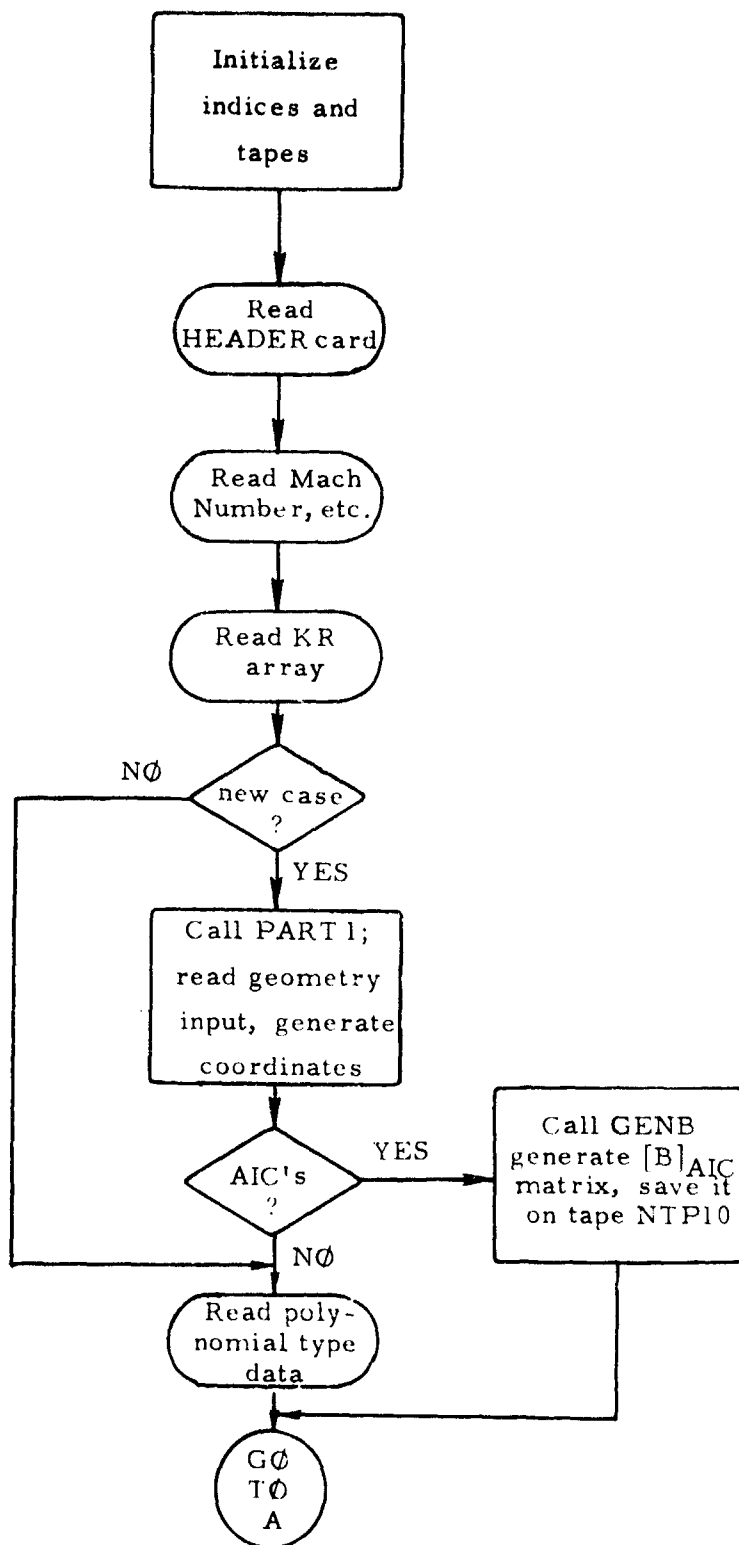
None

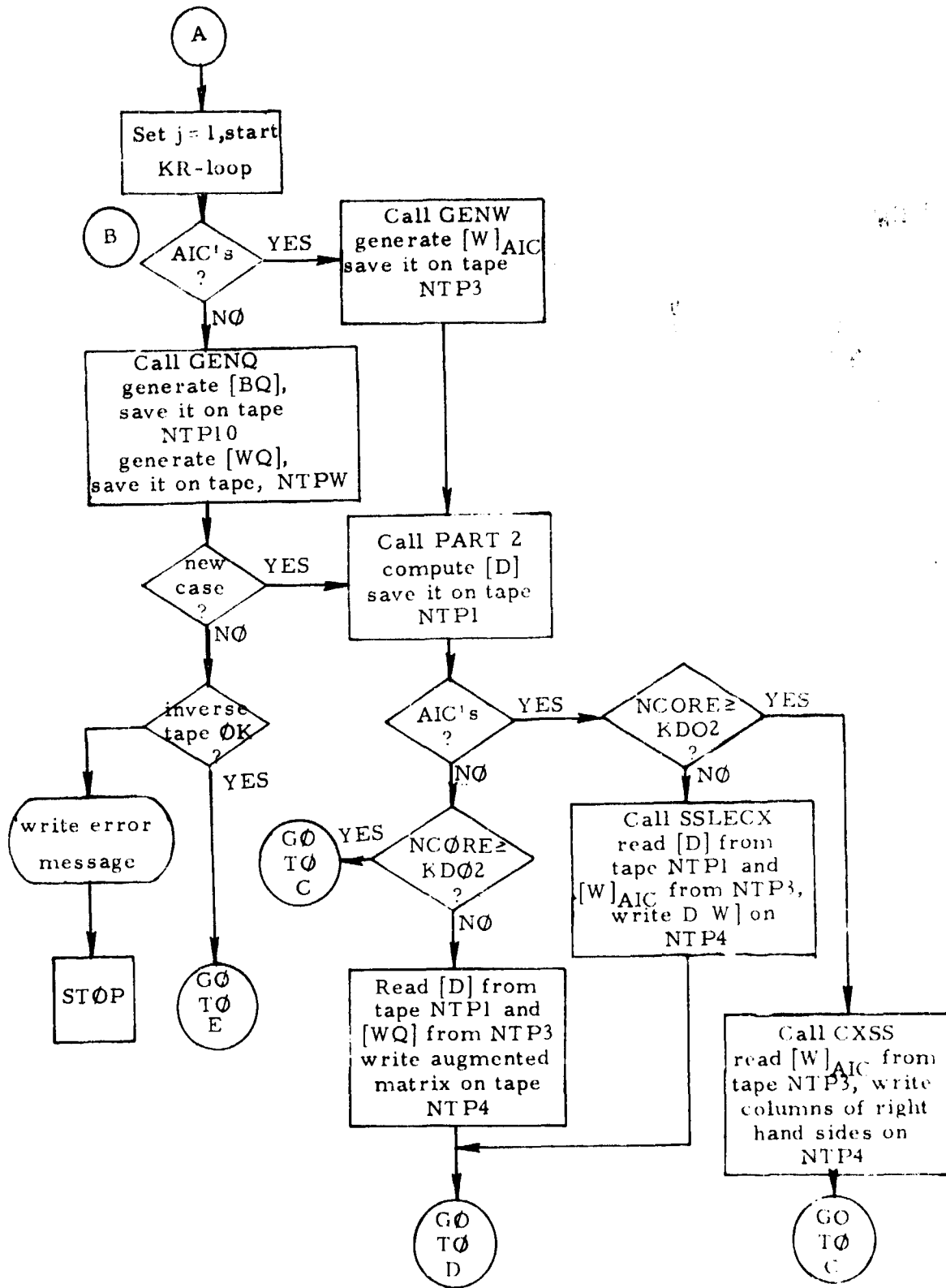
Data in Argument List

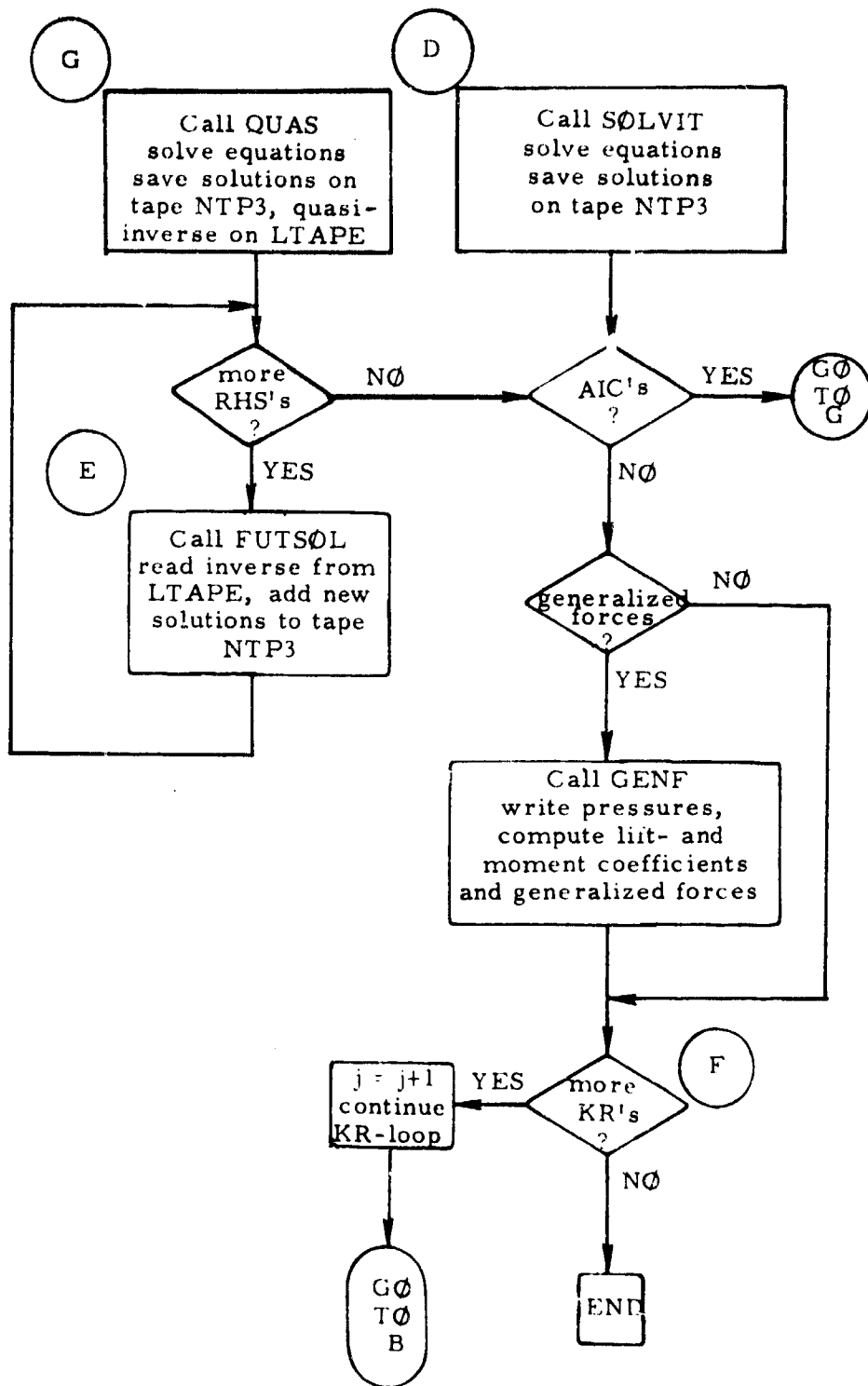
MNEMONICS	SYMBOL	DESCRIPTION
NDEL NB NSTRIP	δ	Symmetry flag Number of bodies } input Number of strips }
NBØX		Total number of boxes on all panels
NTØT	n	NBØX + NBE, where NBE = total number of body elements
NBV		T-tail input
NSV		
NMD	NM, m	Number of modes - input
LIM(50, 3)		See Sec. 2. 5. 3
ACAP, REFARA	A, S	
FL, REFCHD	\bar{c} , $2b_r$	
BZ, REFSPN	s, b/2	
EV(400)	ξ	See Blank Common, Sec. 2. 3. 1
Y(400)	y	
ZZ(400)	z	
SDELX(400)	Δx_s	See Common /XXZ/, Sec. 2. 3. 2
XIJ(50)	x_{le}	
XØC(400)	ξ/c	

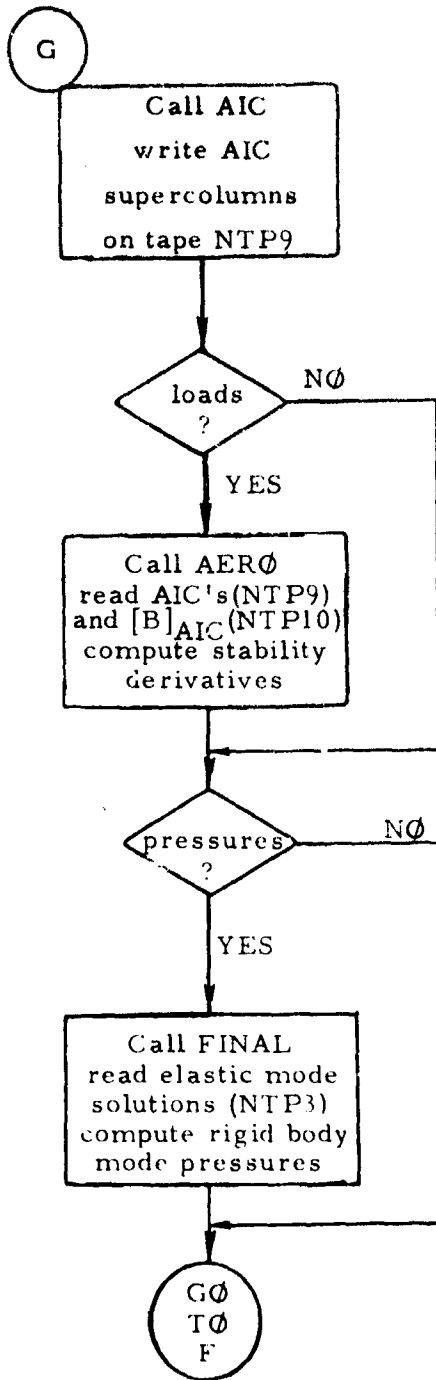
2.6 BLOCK DIAGRAM

The basic block diagram for the computer program H7WC is shown on the next four pages. All logical steps in the "main" part of the program are indicated, and the basic functions of all subroutines that are called from "main" are outlined. A complete description of all subroutines - including those that are not called directly from "main," and therefore are not indicated in the block diagram - can be found in Sections 2.5.1 through 2.5.22.









2.7 PROGRAM LISTING

The following pages (pp. 95 - 143) contain the complete listing of program H7WC. The individual subroutines of the program (including the "main" part of it) are listed in the order of their description in Section 2.5.


```

01/15/71
MAIN - EFN SOURCE STATEMENT - IFNISI -
CALL PARSTRNG(MB)
IF (INT-EG,MPD) GO TO 8000
STOP
453 REWIND LTAPE
REWIND MAT
IMMICH = 1
CALL R-EG, 01 GO TO 8000
IF (R-EG,MPD) IMMICH=2
450 CONTINUE
REWIND MM
8000 GO TO 18000, 9002), IMMICH
CALL TIME PUTSON (MD, R-EG, NE, MM, NO, NAT, NW, LTAPE, RHSTAP, NPR1)
IF (R-EG,MPD) IMMICH=C
9002 CONTINUE
154 CONTINUE (R-EG, 01) GO TO 157
155 CONTINUE
156 CONTINUE
157 CONTINUE
CALL AIC(BR,S, MR,NMF,NBL,JOBN,ICOD,VEL,NGUST, WPR2)
IF (MPR3,EG,01) GO TO 160
CALL AENO(BR,BZ,RP,ACAP,NMF,NSV,N7AM,IPR,MPC,NOELT,NPR3)
CALL FINAL(BR,BZ,ACAP,NSV,NBY,IRI,NTAM,MPC,NOELT,PGANMA)
157 CONTINUE
IF (INT-EG,01) GO TO 160
160 CONTINUE
CALL GENF(MOELT,NB,MSTRIP,NBOI,NTOT,NB,NV,NHO,LIR,ACAP,FL,
82,EV,Y,Z,SDLE,LRJ,ROC,P)
160 CONTINUE
GO TO 100
END
154 154 MAIN1100
163 163 MAIN1170
166 166 MAIN1200
175 175 MAIN1280
176 176 MAIN1310
77 77 MAIN1320
182 182 MAIN1370
187 187 MAIN1390
199 199 MAIN1440
211 211 MAIN1540
212 212 MAIN1570
224 224 MAIN1670
227 227 MAIN1680
275 275 MAIN1700
732 732 MAIN1720

01/15/71
MAIN - EFN SOURCE STATEMENT - IFNISI -
CALL PARSTRNG(MB)
IF (INT-EG,MPD) GO TO 105
IF (MPC,GT,MB) GO TO 142
CALL SAE(LINE,MSI,MD,NE,UST)
GO TO 140
CALL TIME PUTSON (MD, R-EG, NE, MM, NO, NAT, NW, LTAPE, RHSTAP, NPR1)
IF (R-EG,MPD) IMMICH=C
9002 CONTINUE
154 CONTINUE
155 CONTINUE
156 CONTINUE
157 CONTINUE
CALL AIC(BR,S, MR,NMF,NBL,JOBN,ICOD,VEL,NGUST, WPR2)
IF (MPR3,EG,01) GO TO 160
CALL AENO(BR,BZ,RP,ACAP,NMF,NSV,N7AM,IPR,MPC,NOELT,NPR3)
CALL FINAL(BR,BZ,ACAP,NSV,NBY,IRI,NTAM,MPC,NOELT,PGANMA)
157 CONTINUE
IF (INT-EG,01) GO TO 160
160 CONTINUE
CALL GENF(MOELT,NB,MSTRIP,NBOI,NTOT,NB,NV,NHO,LIR,ACAP,FL,
82,EV,Y,Z,SDLE,LRJ,ROC,P)
160 CONTINUE
GO TO 100
END
154 154 MAIN1100
163 163 MAIN1170
166 166 MAIN1200
175 175 MAIN1280
176 176 MAIN1310
77 77 MAIN1320
182 182 MAIN1370
187 187 MAIN1390
199 199 MAIN1440
211 211 MAIN1540
212 212 MAIN1570
224 224 MAIN1670
227 227 MAIN1680
275 275 MAIN1700
732 732 MAIN1720

```


01/15/71

GENM - EFM SOURCE STATEMENT - IFN51 -

```

X5 = XLE + HHE(,5) + CHORD
X6 = XLE + HHE(,6) + CHORD
X7 = XLE + HHE(,7) + CHORD
IF (MPC-445,0) * 21 * (X1 - X3)
BOT1 = (X2 - X1) + (X2 - X3)
BOT2 = (X2 - X1) + (X2 - X3)
GO TO 120
110 BOT1 = 1.
120 BOT2 = 1.
CONTINUE
IC = 0
IF (XISLCT(,1),LE-0,0) GO TO 440
IC = IC + 1
IF (MPC-NE,0) GO TO 290
GO TO 130,150,170,190,210,230,460,4601, J
130 DO 140 L = LIM1,LIM2
IR = L - LIM1 + 1
140 WIR,IC1 = CMPLX(0,0,KRDBR)
150 DO 160 L = LIM1,LIM2
WIR,IC1 = CMPLX(1,0,KRDBR*(KILJ-X1))
160 WIR,IC1 = CMPLX(1,0,KRDBR*(KILJ-X1))
GO TO 440
170 DO 190 L = LIM1,LIM2
IR = L - LIM1 + 1
180 WIR,IC1 = CMPLX(1,0,KRDBR*(KILJ-X3))
GO TO 460
190 DO 200 L = LIM1,LIM2
WIR,IC1 = CMPLX(1,0,KRDBR*(KILJ-X3))
GO TO 460
200 CONTINUE
WIR,IC1 = CMPLX(1,0,KRDBR*(KILJ-X5))
GO TO 460
210 DO 220 L = LIM1,LIM2
IR = L - LIM1 + 1
IF (KILJ-LY,X2) GO TO 220
WIR,IC1 = CMPLX(0,0,KRDBR)
220 CONTINUE
GO TO 460
230 DO 240 L = LIM1,LIM2
IR = L - LIM1 + 1
IF (KILJ-LY,X4) GO TO 240
WIR,IC1 = CMPLX(0,0,KRDBR)
240 CONTINUE
GO TO 460
250 CONTINUE
GO TO 1240,300,340,380,400,420,440,4601, J
260 DO 290 L = LIM1,LIM2
TOP1 = (KILJ-X2) + (KILJ-X3)

```

01/15/71

GENM - EFM SOURCE STATEMENT - IFN51 -

```

BOT = BOT1 + (X5-X2) + (X5-X3)/BOT
IF (XISLCT(,4),NE-0,0,AND,KILJ-GT,X4) GO TO 270
WIR = KRDBR*(TOP1/BOT)
GO TO 280
270 DO 280 L = LIM1,LIM2
DELTA = ((X5-X2) + (X5-X3))/BOT
WIR = DELTA
WIR = KRDBR*(IC + DELTA * (KILJ-X5))
280 WIR,IC = CMPLX(WIR,WIR)
290 CONTINUE
GO TO 480
300 DO 310 L = LIM1,LIM2
IR = L - LIM1 + 1
TOP1 = (KILJ-X1) + (KILJ-X3)
290 TOP1 = (KILJ-X1) + (KILJ-X3)
BOT = BOT2
IF (XISLCT(,4),NE-0,0,AND,KILJ-GT,X4) GO TO 310
WIR = TOP1/BOT
WIR = KRDBR*(TOP1/BOT)
GO TO 320
310 DELTA = ((X5-X2) + (X5-X3))/BOT
WIR = DELTA
WIR = KRDBR*(IC + DELTA*(KILJ-X5))
320 WIR,IC = CMPLX(WIR,WIR)
330 CONTINUE
GO TO 440
340 DO 370 L = LIM1,LIM2
IR = L - LIM1 + 1
TOP1 = (KILJ-X1) + (KILJ-X2)
BOT = BOT1
IF (XISLCT(,4),NE-0,0,AND,KILJ-GT,X4) GO TO 350
WIR = TOP1/BOT
WIR = KRDBR*(TOP1/BOT)
GO TO 360
350 DO 390 L = LIM1,LIM2
DELTA = ((X5-X1) + (X5-X2))/BOT
WIR = DELTA
WIR = KRDBR*(IC + DELTA*(KILJ-X5))
360 WIR,IC = CMPLX(WIR,WIR)
370 CONTINUE
GO TO 460
380 DO 390 L = LIM1,LIM2
IR = L - LIM1 + 1
IF (KILJ-LY,X4) GO TO 390
WIR,IC1 = CMPLX(1,0,KRDBR*(KILJ-X5))
390 CONTINUE
GO TO 460
400 DO 410 L = LIM1,LIM2
IR = L - LIM1 + 1
IF (KILJ-LY,X4) GO TO 410
WIR,IC1 = CMPLX(1,0,KRDBR*(KILJ-X7))
410 CONTINUE
GO TO 440

```



```

1  READ (5,30) M1,N1,MARQ14,1,NB14,1,M12,M02,MARQ1K(1),NB1K(1),
2  M13,M13,MARQ14(2),NS1K(2),M14,M14,MARQ1K(1),NB1K(1),
3  M15,M15,MARQ1K(1),NB1K(1),M16,M16,MARQ1K(1),NB1K(1),
4  M17,M17,MARQ1K(1),NB1K(1)
5  CONTINUE
6  DO 56 I=1,NB14
7  READ (5,30) M1,N1,MARQ14,1,NB14,1,M12,M02,MARQ1K(1),NB1K(1),
8  M13,M13,MARQ14(2),NS1K(2),M14,M14,MARQ1K(1),NB1K(1),
9  M15,M15,MARQ1K(1),NB1K(1),M16,M16,MARQ1K(1),NB1K(1),
10 M17,M17,MARQ1K(1),NB1K(1)
11 WRITE (6,20)
12 CONTINUE
13 DO 58 J=1,NB
14 DO 58 L=1,NB
15 READ (5,30) M1,N1,MARQ14,1,NB14,1,M12,M02,MARQ1K(1),NB1K(1),
16 M13,M13,MARQ14(2),NS1K(2),M14,M14,MARQ1K(1),NB1K(1),
17 M15,M15,MARQ1K(1),NB1K(1),M16,M16,MARQ1K(1),NB1K(1),
18 M17,M17,MARQ1K(1),NB1K(1)
19 IF (M12.EQ.0) GO TO 58
20 DO 57 I=1,NB
21 READ (5,30) M1,N1,MARQ14,1,NB14,1,M12,M02,MARQ1K(1),NB1K(1),
22 M13,M13,MARQ14(2),NS1K(2),M14,M14,MARQ1K(1),NB1K(1),
23 M15,M15,MARQ1K(1),NB1K(1),M16,M16,MARQ1K(1),NB1K(1),
24 M17,M17,MARQ1K(1),NB1K(1)
25 WRITE (6,20)
26 CONTINUE
27 IF (M12.EQ.1) GO TO 57
28 CONTINUE
29 IF (M12.EQ.1) GO TO 57
30 CONTINUE
31 IF (M12.EQ.1) GO TO 57
32 CONTINUE
33 IF (M12.EQ.1) GO TO 57
34 CONTINUE
35 IF (M12.EQ.1) GO TO 57
36 CONTINUE
37 IF (M12.EQ.1) GO TO 57
38 CONTINUE
39 IF (M12.EQ.1) GO TO 57
40 CONTINUE
41 IF (M12.EQ.1) GO TO 57
42 CONTINUE
43 IF (M12.EQ.1) GO TO 57
44 CONTINUE
45 IF (M12.EQ.1) GO TO 57
46 CONTINUE
47 IF (M12.EQ.1) GO TO 57
48 CONTINUE
49 IF (M12.EQ.1) GO TO 57
50 CONTINUE
51 IF (M12.EQ.1) GO TO 57
52 CONTINUE
53 IF (M12.EQ.1) GO TO 57
54 CONTINUE
55 IF (M12.EQ.1) GO TO 57
56 CONTINUE
57 IF (M12.EQ.1) GO TO 57
58 CONTINUE
59 IF (M12.EQ.1) GO TO 57
60 CONTINUE
61 IF (M12.EQ.1) GO TO 57
62 CONTINUE
63 IF (M12.EQ.1) GO TO 57
64 CONTINUE
65 IF (M12.EQ.1) GO TO 57
66 CONTINUE
67 IF (M12.EQ.1) GO TO 57
68 CONTINUE
69 IF (M12.EQ.1) GO TO 57
70 CONTINUE
71 IF (M12.EQ.1) GO TO 57
72 CONTINUE
73 IF (M12.EQ.1) GO TO 57
74 CONTINUE
75 IF (M12.EQ.1) GO TO 57
76 CONTINUE
77 IF (M12.EQ.1) GO TO 57
78 CONTINUE
79 IF (M12.EQ.1) GO TO 57
80 CONTINUE
81 IF (M12.EQ.1) GO TO 57
82 CONTINUE
83 IF (M12.EQ.1) GO TO 57
84 CONTINUE
85 IF (M12.EQ.1) GO TO 57
86 CONTINUE
87 IF (M12.EQ.1) GO TO 57
88 CONTINUE
89 IF (M12.EQ.1) GO TO 57
90 CONTINUE
91 IF (M12.EQ.1) GO TO 57
92 CONTINUE
93 IF (M12.EQ.1) GO TO 57
94 CONTINUE
95 IF (M12.EQ.1) GO TO 57
96 CONTINUE
97 IF (M12.EQ.1) GO TO 57
98 CONTINUE
99 IF (M12.EQ.1) GO TO 57
100 CONTINUE

```

```

81 CONTINUE
82 GO TO 100
83 CONTINUE
84 CONTINUE
85 CONTINUE
86 CONTINUE
87 CONTINUE
88 CONTINUE
89 CONTINUE
90 CONTINUE
91 CONTINUE
92 CONTINUE
93 CONTINUE
94 CONTINUE
95 CONTINUE
96 CONTINUE
97 CONTINUE
98 CONTINUE
99 CONTINUE
100 CONTINUE
101 CONTINUE
102 CONTINUE
103 CONTINUE
104 CONTINUE
105 CONTINUE
106 CONTINUE
107 CONTINUE
108 CONTINUE
109 CONTINUE
110 CONTINUE
111 CONTINUE
112 CONTINUE
113 CONTINUE
114 CONTINUE
115 CONTINUE
116 CONTINUE
117 CONTINUE
118 CONTINUE
119 CONTINUE
120 CONTINUE
121 CONTINUE
122 CONTINUE
123 CONTINUE
124 CONTINUE
125 CONTINUE
126 CONTINUE
127 CONTINUE
128 CONTINUE
129 CONTINUE
130 CONTINUE
131 CONTINUE
132 CONTINUE
133 CONTINUE
134 CONTINUE
135 CONTINUE
136 CONTINUE
137 CONTINUE
138 CONTINUE
139 CONTINUE
140 CONTINUE
141 CONTINUE
142 CONTINUE
143 CONTINUE
144 CONTINUE
145 CONTINUE
146 CONTINUE
147 CONTINUE
148 CONTINUE
149 CONTINUE
150 CONTINUE
151 CONTINUE
152 CONTINUE
153 CONTINUE
154 CONTINUE
155 CONTINUE
156 CONTINUE
157 CONTINUE
158 CONTINUE
159 CONTINUE
160 CONTINUE
161 CONTINUE
162 CONTINUE
163 CONTINUE
164 CONTINUE
165 CONTINUE
166 CONTINUE
167 CONTINUE
168 CONTINUE
169 CONTINUE
170 CONTINUE
171 CONTINUE
172 CONTINUE
173 CONTINUE
174 CONTINUE
175 CONTINUE
176 CONTINUE
177 CONTINUE
178 CONTINUE
179 CONTINUE
180 CONTINUE
181 CONTINUE
182 CONTINUE
183 CONTINUE
184 CONTINUE
185 CONTINUE
186 CONTINUE
187 CONTINUE
188 CONTINUE
189 CONTINUE
190 CONTINUE
191 CONTINUE
192 CONTINUE
193 CONTINUE
194 CONTINUE
195 CONTINUE
196 CONTINUE
197 CONTINUE
198 CONTINUE
199 CONTINUE
200 CONTINUE

```

Best Available Copy

```

140 CONTINUE
150 CONTINUE
160 CONTINUE
170 CONTINUE
180 CONTINUE
190 CONTINUE
200 CONTINUE
210 CONTINUE
220 CONTINUE
230 CONTINUE
240 CONTINUE
250 CONTINUE
260 CONTINUE
270 CONTINUE
280 CONTINUE
290 CONTINUE
300 CONTINUE
310 CONTINUE
320 CONTINUE
330 CONTINUE
340 CONTINUE
350 CONTINUE
360 CONTINUE
370 CONTINUE
380 CONTINUE
390 CONTINUE
400 CONTINUE
410 CONTINUE
420 CONTINUE
430 CONTINUE
440 CONTINUE
450 CONTINUE
460 CONTINUE
470 CONTINUE
480 CONTINUE
490 CONTINUE
500 CONTINUE
510 CONTINUE
520 CONTINUE
530 CONTINUE
540 CONTINUE
550 CONTINUE
560 CONTINUE
570 CONTINUE
580 CONTINUE
590 CONTINUE
600 CONTINUE
610 CONTINUE
620 CONTINUE
630 CONTINUE
640 CONTINUE
650 CONTINUE
660 CONTINUE
670 CONTINUE
680 CONTINUE
690 CONTINUE
700 CONTINUE
710 CONTINUE
720 CONTINUE
730 CONTINUE
740 CONTINUE
750 CONTINUE
760 CONTINUE
770 CONTINUE
780 CONTINUE
790 CONTINUE
800 CONTINUE
810 CONTINUE
820 CONTINUE
830 CONTINUE
840 CONTINUE
850 CONTINUE
860 CONTINUE
870 CONTINUE
880 CONTINUE
890 CONTINUE
900 CONTINUE
910 CONTINUE
920 CONTINUE
930 CONTINUE
940 CONTINUE
950 CONTINUE
960 CONTINUE
970 CONTINUE
980 CONTINUE
990 CONTINUE
1000 CONTINUE

```

```

AUG1590
AUG1591
AUG1592
AUG1593
AUG1594
AUG1595
AUG1596
AUG1597
AUG1598
AUG1599
AUG1600
AUG1601
AUG1602
AUG1603
AUG1604
AUG1605
AUG1606
AUG1607
AUG1608
AUG1609
AUG1610
AUG1611
AUG1612
AUG1613
AUG1614
AUG1615
AUG1616
AUG1617
AUG1618
AUG1619
AUG1620
AUG1621
AUG1622
AUG1623
AUG1624
AUG1625
AUG1626
AUG1627
AUG1628
AUG1629
AUG1630
AUG1631
AUG1632
AUG1633
AUG1634
AUG1635
AUG1636
AUG1637
AUG1638
AUG1639
AUG1640
AUG1641
AUG1642
AUG1643
AUG1644
AUG1645
AUG1646
AUG1647
AUG1648
AUG1649
AUG1650
AUG1651
AUG1652
AUG1653
AUG1654
AUG1655
AUG1656
AUG1657
AUG1658
AUG1659
AUG1660
AUG1661
AUG1662
AUG1663
AUG1664
AUG1665
AUG1666
AUG1667
AUG1668
AUG1669
AUG1670
AUG1671
AUG1672
AUG1673
AUG1674
AUG1675
AUG1676
AUG1677
AUG1678
AUG1679
AUG1680
AUG1681
AUG1682
AUG1683
AUG1684
AUG1685
AUG1686
AUG1687
AUG1688
AUG1689
AUG1690
AUG1691
AUG1692
AUG1693
AUG1694
AUG1695
AUG1696
AUG1697
AUG1698
AUG1699
AUG1700
AUG1701
AUG1702
AUG1703
AUG1704
AUG1705
AUG1706
AUG1707
AUG1708
AUG1709
AUG1710
AUG1711
AUG1712
AUG1713
AUG1714
AUG1715
AUG1716
AUG1717
AUG1718
AUG1719
AUG1720
AUG1721
AUG1722
AUG1723
AUG1724
AUG1725
AUG1726
AUG1727
AUG1728
AUG1729
AUG1730
AUG1731
AUG1732
AUG1733
AUG1734
AUG1735
AUG1736
AUG1737
AUG1738
AUG1739
AUG1740
AUG1741
AUG1742
AUG1743
AUG1744
AUG1745
AUG1746
AUG1747
AUG1748
AUG1749
AUG1750
AUG1751
AUG1752
AUG1753
AUG1754
AUG1755
AUG1756
AUG1757
AUG1758
AUG1759
AUG1760
AUG1761
AUG1762
AUG1763
AUG1764
AUG1765
AUG1766
AUG1767
AUG1768
AUG1769
AUG1770
AUG1771
AUG1772
AUG1773
AUG1774
AUG1775
AUG1776
AUG1777
AUG1778
AUG1779
AUG1780
AUG1781
AUG1782
AUG1783
AUG1784
AUG1785
AUG1786
AUG1787
AUG1788
AUG1789
AUG1790
AUG1791
AUG1792
AUG1793
AUG1794
AUG1795
AUG1796
AUG1797
AUG1798
AUG1799
AUG1800

```


PART2 - EFM SOURCE STATEMENT - (PWL1)

```

CALL      = DL SMPCC=CL,CL,TL,SGS,CUS,SGM,COR,RO,TO,ZO,EE,UI,J,BETA,CV,PREZLKO
CONTIME  = DL *DIJ *IP(EDATEMULTI)
330      IF JSPEC=EO,0) GO TO 350
340      WABLE = JSPEC
350      GMSIG = GRATIG)
SGS      = SINGGMSIG)
411      = ZL(10) *ZL(10)
422      = ZL(10) *ZL(10)
JSPEC    = 0
MULT     = JSPEC)
COUNT  = 10
360      CONTIME
DELR    = SOELR *FUELE
DELT    = SOELI *SOFLE *PODEL
DIJ     = SVELI *SVELI *PODEL
AMBITA  = DIJ * DELR
DPMELI  = DM * DL *PODEL
IR      = IR * 1
GMSIG   = GRATIG)
GMS     = GMSIG)
SGS     = SINGGMSIG)
370      = SINGGMSIG)
IF (RO,CF,MC,MS) GO TO 380
10      = 10 * 1
IF 110,CF,MC,MS) 10 = 1
380      = 10 * 1
390      = 10 * 1
400      = 10 * 1
410      = 10 * 1
420      = 10 * 1
430      = 10 * 1
440      = 10 * 1
450      = 10 * 1
460      = 10 * 1
470      = 10 * 1
480      = 10 * 1
490      = 10 * 1
500      = 10 * 1
510      = 10 * 1
520      = 10 * 1
530      = 10 * 1
540      = 10 * 1
550      = 10 * 1
560      = 10 * 1
570      = 10 * 1
580      = 10 * 1
590      = 10 * 1
600      = 10 * 1
610      = 10 * 1
620      = 10 * 1
630      = 10 * 1
640      = 10 * 1
650      = 10 * 1
660      = 10 * 1
670      = 10 * 1
680      = 10 * 1
690      = 10 * 1
700      = 10 * 1
710      = 10 * 1
720      = 10 * 1
730      = 10 * 1
740      = 10 * 1
750      = 10 * 1
760      = 10 * 1
770      = 10 * 1
780      = 10 * 1
790      = 10 * 1
800      = 10 * 1
810      = 10 * 1
820      = 10 * 1
830      = 10 * 1
840      = 10 * 1
850      = 10 * 1
860      = 10 * 1
870      = 10 * 1
880      = 10 * 1
890      = 10 * 1
900      = 10 * 1
910      = 10 * 1
920      = 10 * 1
930      = 10 * 1
940      = 10 * 1
950      = 10 * 1
960      = 10 * 1
970      = 10 * 1
980      = 10 * 1
990      = 10 * 1
1000     = 10 * 1

```

INCRD - 14N SOURCE STATEMENT - (PWL1)

```

SUBROUTINE INCRD(AN,AY,AZ,AR,AV,AVI,AVI,ARZ,AYZ,AZZ,GAMS,GAMSIG,AWM,INCRD010
CONTIME  = DL *DIJ *IP(EDATEMULTI)
330      IF JSPEC=EO,0) GO TO 350
340      WABLE = JSPEC
350      GMSIG = GRATIG)
SGS      = SINGGMSIG)
411      = ZL(10) *ZL(10)
422      = ZL(10) *ZL(10)
JSPEC    = 0
MULT     = JSPEC)
COUNT  = 10
360      CONTIME
DELR    = SOELR *FUELE
DELT    = SOELI *SOFLE *PODEL
DIJ     = SVELI *SVELI *PODEL
AMBITA  = DIJ * DELR
DPMELI  = DM * DL *PODEL
IR      = IR * 1
GMSIG   = GRATIG)
GMS     = GMSIG)
SGS     = SINGGMSIG)
370      = SINGGMSIG)
IF (RO,CF,MC,MS) GO TO 380
10      = 10 * 1
IF 110,CF,MC,MS) 10 = 1
380      = 10 * 1
390      = 10 * 1
400      = 10 * 1
410      = 10 * 1
420      = 10 * 1
430      = 10 * 1
440      = 10 * 1
450      = 10 * 1
460      = 10 * 1
470      = 10 * 1
480      = 10 * 1
490      = 10 * 1
500      = 10 * 1
510      = 10 * 1
520      = 10 * 1
530      = 10 * 1
540      = 10 * 1
550      = 10 * 1
560      = 10 * 1
570      = 10 * 1
580      = 10 * 1
590      = 10 * 1
600      = 10 * 1
610      = 10 * 1
620      = 10 * 1
630      = 10 * 1
640      = 10 * 1
650      = 10 * 1
660      = 10 * 1
670      = 10 * 1
680      = 10 * 1
690      = 10 * 1
700      = 10 * 1
710      = 10 * 1
720      = 10 * 1
730      = 10 * 1
740      = 10 * 1
750      = 10 * 1
760      = 10 * 1
770      = 10 * 1
780      = 10 * 1
790      = 10 * 1
800      = 10 * 1
810      = 10 * 1
820      = 10 * 1
830      = 10 * 1
840      = 10 * 1
850      = 10 * 1
860      = 10 * 1
870      = 10 * 1
880      = 10 * 1
890      = 10 * 1
900      = 10 * 1
910      = 10 * 1
920      = 10 * 1
930      = 10 * 1
940      = 10 * 1
950      = 10 * 1
960      = 10 * 1
970      = 10 * 1
980      = 10 * 1
990      = 10 * 1
1000     = 10 * 1

```


1947-1948
1948-1949
1949-1950
1950-1951
1951-1952
1952-1953
1953-1954
1954-1955
1955-1956
1956-1957
1957-1958
1958-1959
1959-1960
1960-1961
1961-1962
1962-1963
1963-1964
1964-1965
1965-1966
1966-1967
1967-1968
1968-1969
1969-1970
1970-1971
1971-1972
1972-1973
1973-1974
1974-1975
1975-1976
1976-1977
1977-1978
1978-1979
1979-1980
1980-1981
1981-1982
1982-1983
1983-1984
1984-1985
1985-1986
1986-1987
1987-1988
1988-1989
1989-1990
1990-1991
1991-1992
1992-1993
1993-1994
1994-1995
1995-1996
1996-1997
1997-1998
1998-1999
1999-2000
2000-2001
2001-2002
2002-2003
2003-2004
2004-2005
2005-2006
2006-2007
2007-2008
2008-2009
2009-2010
2010-2011
2011-2012
2012-2013
2013-2014
2014-2015
2015-2016
2016-2017
2017-2018
2018-2019
2019-2020
2020-2021
2021-2022
2022-2023
2023-2024
2024-2025

1947-1948
1948-1949
1949-1950
1950-1951
1951-1952
1952-1953
1953-1954
1954-1955
1955-1956
1956-1957
1957-1958
1958-1959
1959-1960
1960-1961
1961-1962
1962-1963
1963-1964
1964-1965
1965-1966
1966-1967
1967-1968
1968-1969
1969-1970
1970-1971
1971-1972
1972-1973
1973-1974
1974-1975
1975-1976
1976-1977
1977-1978
1978-1979
1979-1980
1980-1981
1981-1982
1982-1983
1983-1984
1984-1985
1985-1986
1986-1987
1987-1988
1988-1989
1989-1990
1990-1991
1991-1992
1992-1993
1993-1994
1994-1995
1995-1996
1996-1997
1997-1998
1998-1999
1999-2000
2000-2001
2001-2002
2002-2003
2003-2004
2004-2005
2005-2006
2006-2007
2007-2008
2008-2009
2009-2010
2010-2011
2011-2012
2012-2013
2013-2014
2014-2015
2015-2016
2016-2017
2017-2018
2018-2019
2019-2020
2020-2021
2021-2022
2022-2023
2023-2024
2024-2025

Best Available Copy

01/15/71

```

SOLVET - EFN SOURCE STATEMENT - IPHES) -
DO 10 NT = 1, NP
  READ (UNIT) (A1(I), I = 1, NS, NT)
  DO 20 NP = 1, N
    NS = NS + 1
    IF (NS .EQ. N) GO TO 10
  -- READ ANOTHER ROW
  DO 30 I = 1, NP
    READ (UNIT) (A1(I), I = 1, NS, A(N))
  -- MODIFY THIS ROW BY THE 'TRAPEZOIDAL' ARRAY
  NT = 1
  DO 40 I = 1, NS
    A(I) = A(I) - A(N) / A(NT)
  TEMP = REAL(A(N))
  DENOM = - (REAL(A(NT))) * 2 + ATNAG(A(NT)) * 2
  RAL1(N) = (REAL(A(NT))) / DENOM
  RAL2(N) = (ATNAG(A(N))) * REAL(A(NT)) - TEMP
  RAL3(N) = (ATNAG(A(NT))) / DENOM
  DO 65 NN = NP, NORE
    NP = NP + 1
  65 A(N) = A(N) + A(N) * A(N)
  RAL1(N) = RAL1(N) / REAL(A(N)) + REAL(A(N))
  RAL2(N) = RAL2(N) + REAL(A(N)) * ATNAG(A(N))
  RAL3(N) = RAL3(N) + REAL(A(N))
  NP = NP
  -- WRITE THE MODIFIED ROW ON TAPE
  DO WRITE (UNIT) (A1(NT), NT = NN, NORE)
  REWIND UNIT
  REWIND UNIT
  -- SWITCH THE TAPES
  NT = NT + 1
  IF (NT .EQ. N) GO TO 10
  RE-CALCULATE ROW LENGTH AND LOOP BACK

```

```

SOLVIT - EFN SOURCE STATEMENT - IPHES) -
  MEL = 1
  NN = MEL
  GO TO 10
  -- REWIND ALL TAPES
  NO REWIND UNIT
  REWIND UNIT
  REWIND UNIT
  -- CONDENSE THE MATRIX
  NN = MEL
  NL = MEL
  IF (NL .EQ. 1) GO TO 105
  NT = MEL
  DO 100 I = 1, N
    NS = NS + REAL(1)
    NT = NT + MEL
    GO TO 105
  100 NL = NL + 1
  105 NL = NORE - K + 1
  -- THERE, NOW WE CAN START THE BACK-SOLUTION
  * * * NOTE: THE FIRST AVAILABLE LOCATION FOR THE SOLUTIONS IS A(N)
  NNER = N
  MEL = NNR
  LAST = K .ED. N
  NPASS = 0
  -- SOLVE FOR THE ANSWERS CORRESPONDING TO 'A' ROWS
  110 KRI = K - 1
  KPI = K + 1
  NS = NL
  NPASS = NPASS + 1
  DO 130 NN = 1, N
    MF = NS + NN
    A(NF) = A(NF) / A(NS)
  TEMP = REAL(A(NS))
  DENOM = REAL(A(NS)) * 2 + ATNAG(A(NS)) * 2
  RAL1(NF) = (TEMP) * REAL(A(NS)) + ATNAG(A(NF))
  RAL2(NF) = (ATNAG(A(NS)) * REAL(A(NS)) - TEMP)
  RAL3(NF) = (ATNAG(A(NS)) / DENOM
  NT = NS
  IF (KRI .EQ. 0) GO TO 130
  DO 125 I = 1, KRI
    MF = MF - I - 1
    NT = NT - I - 1

```

142
143
144

94
107
146
149
156

SOLV1120
SOLV1130
SOLV1140
SOLV1150
SOLV1160
SOLV1170
SOLV1180
SOLV1190
SOLV1200
SOLV1210
SOLV1220
SOLV1230
SOLV1240
SOLV1250
SOLV1260
SOLV1270
SOLV1280
SOLV1290
SOLV1300
SOLV1310
SOLV1320
SOLV1330
SOLV1340
SOLV1350
SOLV1360
SOLV1370
SOLV1380
SOLV1390
SOLV1400
SOLV1410
SOLV1420
SOLV1430
SOLV1440
SOLV1450
SOLV1460
SOLV1470
SOLV1480
SOLV1490
SOLV1500
SOLV1510
SOLV1520
SOLV1530
SOLV1540
SOLV1550
SOLV1560
SOLV1570
SOLV1580
SOLV1590
SOLV1600
SOLV1610
SOLV1620
SOLV1630
SOLV1640
SOLV1650
SOLV1660
SOLV1670

```

01/15/71
SOLVIT - EFM SOURCE STATEMENT - (FMS) -
C
C SUM = 0.0
C SUM1 = 0.0
C SUM2 = 0.0
C
C N2 = NP * 16
C DO 120 I1 = 1, 16
C   NP = NT * I1
C   NP = NP * I2 - I0
C 120 SUM = SUM + AIN(I1) * AIN(I2)
C
C SUM = SUM + REAL(AIN(I1)) * REAL(AIN(I2)) - AIMAG(AIN(I1)) * AIMAG(AIN(I2))
C
C 120 SUM1 = SUM1 + REAL(AIN(I1)) * REAL(AIN(I2)) + AIMAG(AIN(I1)) * AIMAG(AIN(I2))
C
C 125 AIN(I1) = (AIN(I1) - SUM) / AIN(I1)
C
C DENOM = REAL(AIN(I1)) * 2 + AIMAG(AIN(I1)) * 2
C
C TEMP1 = REAL(AIN(I1)) - SUM
C TEMP2 = AIMAG(AIN(I1)) - SUM1
C
C 125 REAL(I1) = (TEMP1 * REAL(AIN(I1)) + TEMP2 * AIMAG(AIN(I1))) / DENOM
C 125 AIMAG(I1) = (TEMP1 * AIMAG(AIN(I1)) - TEMP2 * REAL(AIN(I1))) / DENOM
C 130 CONTINUE
C
C -- MOVE THE SOLUTIONS TO CONTIGUOUS LOCATIONS STARTING AT AIN1
C
C N1 = 4096 + 1
C DO 140 NN = 1, N1
C   N2 = N1 - 1
C   N3 = N1 - 2
C   N4 = N1 - 3
C 135 AIN(N1) = AIN(N1)
C 140 N1 = N1 - NN
C
C -- WRITE THE SOLUTIONS ON TAPE
C
C WRITE (UNIT, *)
C N5 = N1 - 1
C DO 145 NN = 1, N1
C   WRITE (UNIT, *) AIN(N1), I0 = NT, WORK, N1
C 145 N1 = N1 - NN
C
C -- FIRST 16 PASSES OF THE FIRST PASS
C
C 15 CLASS UP TO 700
C
C -- THE FIRST FOUR PASSES OF THE TRIANGULAR MATRIX TO REFLECT THE EFFECT OF
C THE SOLUTIONS OBTAINED IN PASSES 160-167

```

```

01/15/71
SOLVIT - EFM SOURCE STATEMENT - (FMS) -
C
C -- NOTE...LOCATIONS A(1) TO A(N1-1) ARE NOW FREE TO USE
C -- CALCULATE THE NEXT VALUES OF *NEL* AND *NREM*
C
C KOLD = K
C NEL = NEL - K
C NREM = NREM - K
C
C -- NOW APPLY THE INCREDIBLE FORMULA FOR THE NEW *K*
C
C K = (1.4 * M - 1) / 2 * ((K(SORTIO.25) * FLOAT(6 * M * 2) * M *
C 160M) * NREM - KOLD)
C IF (K - LT * NREM) GO TO 150
C LAST = *TRUE*
C NREM = 1
C K = NREM
C 150 NT = KOLD * 1
C
C -- READ IN THE ROWS TO BE MODIFIED
C
C DO 160 I1 = 1, NREM
C   N1 = NT * I1
C   N2 = N1 - 1
C   N3 = N1 - 2
C   N4 = N1 - 3
C   N5 = N1 - 4
C 160 READ (RT, *) N1, (A(I1)), I0 = N5, NT)
C
C DO 170 NN = 1, M
C   N2 = N1 - NN
C   N3 = N1 - NN
C   N4 = N1 - NN
C   N5 = N1 - NN
C
C SUM = 0.0
C SUMR = 0.0
C SUMI = 0.0
C
C DO 165 I0 = 1, *NLD
C   SUM = SUM + AIN(I1) * AIN(I2)
C   SUMR = SUMR + REAL(AIN(I1)) * REAL(AIN(I2)) - AIMAG(AIN(I1)) * AIMAG(AIN(I2))
C   SUMI = SUMI + REAL(AIN(I1)) * AIMAG(AIN(I2)) - AIMAG(AIN(I1)) * REAL(AIN(I2))
C
C N2 = N2 + 1
C N3 = N3 + 1
C N4 = N4 + 1
C N5 = N5 + 1
C 170 AIN(I1) = AIN(I1) - SUM
C 170 AIN(I2) = AIN(I2) - SUM

```

```

01/15/71
SOLVIT - EFM SOURCE STATEMENT - (FMS) -
C
C -- NOTE...LOCATIONS A(1) TO A(N1-1) ARE NOW FREE TO USE
C -- CALCULATE THE NEXT VALUES OF *NEL* AND *NREM*
C
C KOLD = K
C NEL = NEL - K
C NREM = NREM - K
C
C -- NOW APPLY THE INCREDIBLE FORMULA FOR THE NEW *K*
C
C K = (1.4 * M - 1) / 2 * ((K(SORTIO.25) * FLOAT(6 * M * 2) * M *
C 160M) * NREM - KOLD)
C IF (K - LT * NREM) GO TO 150
C LAST = *TRUE*
C NREM = 1
C K = NREM
C 150 NT = KOLD * 1
C
C -- READ IN THE ROWS TO BE MODIFIED
C
C DO 160 I1 = 1, NREM
C   N1 = NT * I1
C   N2 = N1 - 1
C   N3 = N1 - 2
C   N4 = N1 - 3
C   N5 = N1 - 4
C 160 READ (RT, *) N1, (A(I1)), I0 = N5, NT)
C
C DO 170 NN = 1, M
C   N2 = N1 - NN
C   N3 = N1 - NN
C   N4 = N1 - NN
C   N5 = N1 - NN
C
C SUM = 0.0
C SUMR = 0.0
C SUMI = 0.0
C
C DO 165 I0 = 1, *NLD
C   SUM = SUM + AIN(I1) * AIN(I2)
C   SUMR = SUMR + REAL(AIN(I1)) * REAL(AIN(I2)) - AIMAG(AIN(I1)) * AIMAG(AIN(I2))
C   SUMI = SUMI + REAL(AIN(I1)) * AIMAG(AIN(I2)) - AIMAG(AIN(I1)) * REAL(AIN(I2))
C
C N2 = N2 + 1
C N3 = N3 + 1
C N4 = N4 + 1
C N5 = N5 + 1
C 170 AIN(I1) = AIN(I1) - SUM
C 170 AIN(I2) = AIN(I2) - SUM

```

01/15/71
 SOLVIT - EPH SOURCE STATEMENT - (PHN51) -
 NS - NT - 1
 NT - NT - 1
 IF IMPR1.EG.OI GO TO 230
 WRITE (6,3) I, I(1), M, - NS, NT
 WRITE (6,4) I(1), M, - NS, NT
 230 WRITE (6,1) (I(1), M, - NS, NT)
 C
 300 FORMAT (A30R15, 2F15, 12H-NETRIA WITH 10, 35H-NIGHT SIDES
 11, SOLVED DIRECTLY IN P8.3, 4H-MINUTS, 1
 12, I, I(1), M, - NS, NT)
 END

01/15/71
 SOLVIT - EPH SOURCE STATEMENT - (PHN51) -
 NS - NT - 1
 NT - NT - 1
 IF IMPR1.EG.OI GO TO 230
 WRITE (6,3) I, I(1), M, - NS, NT
 WRITE (6,4) I(1), M, - NS, NT
 230 WRITE (6,1) (I(1), M, - NS, NT)
 C
 300 FORMAT (A30R15, 2F15, 12H-NETRIA WITH 10, 35H-NIGHT SIDES
 11, SOLVED DIRECTLY IN P8.3, 4H-MINUTS, 1
 12, I, I(1), M, - NS, NT)
 END

01/15/71
 SOLVIT - EPH SOURCE STATEMENT - (PHN51) -
 NS - NT - 1
 NT - NT - 1
 IF IMPR1.EG.OI GO TO 230
 WRITE (6,3) I, I(1), M, - NS, NT
 WRITE (6,4) I(1), M, - NS, NT
 230 WRITE (6,1) (I(1), M, - NS, NT)
 C
 300 FORMAT (A30R15, 2F15, 12H-NETRIA WITH 10, 35H-NIGHT SIDES
 11, SOLVED DIRECTLY IN P8.3, 4H-MINUTS, 1
 12, I, I(1), M, - NS, NT)
 END

01/15/71
 SOLVIT - EPH SOURCE STATEMENT - (PHN51) -
 NS - NT - 1
 NT - NT - 1
 IF IMPR1.EG.OI GO TO 230
 WRITE (6,3) I, I(1), M, - NS, NT
 WRITE (6,4) I(1), M, - NS, NT
 230 WRITE (6,1) (I(1), M, - NS, NT)
 C
 300 FORMAT (A30R15, 2F15, 12H-NETRIA WITH 10, 35H-NIGHT SIDES
 11, SOLVED DIRECTLY IN P8.3, 4H-MINUTS, 1
 12, I, I(1), M, - NS, NT)
 END

01/15/71
 SOLVIT - EPH SOURCE STATEMENT - (PHN51) -
 NS - NT - 1
 NT - NT - 1
 IF IMPR1.EG.OI GO TO 230
 WRITE (6,3) I, I(1), M, - NS, NT
 WRITE (6,4) I(1), M, - NS, NT
 230 WRITE (6,1) (I(1), M, - NS, NT)
 C
 300 FORMAT (A30R15, 2F15, 12H-NETRIA WITH 10, 35H-NIGHT SIDES
 11, SOLVED DIRECTLY IN P8.3, 4H-MINUTS, 1
 12, I, I(1), M, - NS, NT)
 END

01/15/71

```

AIC - EFN SOURCE STATEMENT - (FMIS) -
CH11J,JK3 = CH11J,JK3+0(11),L1)D1JRMILA,JAPFACTOR
440 CONTINUE (G,0) GO TO 530
IF (MUST,STRIP) GO TO 530
DO 520 I1 = 1,NDOP1
G(I1) = 0.
DO 520 L = 1,NDOP2
G(L) = LING * L
G(1) = 50.11443E-04*VEL*0(11),L1)PCGUST(LG1+G(I1))
530 CONTINUE
LING = LING + NUOP2
530 CONTINUE
IF (MPRZ,EG,0) GO TO 537
IF (L,EG,1) WRITE (6,5) NSTRIP
DO 535 JM = 1,NDOP
DO 535 JN = 1,NDOP
WRITE (6,15) K, KB, JM, (CH1JM,JMS,KB), JMS-1,NDOP(1)
535 CONTINUE
537 CONTINUE
540 CONTINUE
RETURN
END

```

01/15/71

```

AIC - EFN SOURCE STATEMENT - (FMIS) -
SUMMARY AIC(S),S,HR,MF,MEL,JOBN,ICDO,VEL,NGUST,MPRZ)
CONTINUE (G,0) GO TO 530
IF (MUST,STRIP) GO TO 530
DO 520 I1 = 1,NDOP1
G(I1) = 0.
DO 520 L = 1,NDOP2
G(L) = LING * L
G(1) = 50.11443E-04*VEL*0(11),L1)PCGUST(LG1+G(I1))
530 CONTINUE
LING = LING + NUOP2
530 CONTINUE
IF (MPRZ,EG,0) GO TO 537
IF (L,EG,1) WRITE (6,5) NSTRIP
DO 535 JM = 1,NDOP
DO 535 JN = 1,NDOP
WRITE (6,15) K, KB, JM, (CH1JM,JMS,KB), JMS-1,NDOP(1)
535 CONTINUE
537 CONTINUE
540 CONTINUE
RETURN
END

```

01/15/71

```

AIC - EFN SOURCE STATEMENT - (FMIS) -
SUMMARY AIC(S),S,HR,MF,MEL,JOBN,ICDO,VEL,NGUST,MPRZ)
CONTINUE (G,0) GO TO 530
IF (MUST,STRIP) GO TO 530
DO 520 I1 = 1,NDOP1
G(I1) = 0.
DO 520 L = 1,NDOP2
G(L) = LING * L
G(1) = 50.11443E-04*VEL*0(11),L1)PCGUST(LG1+G(I1))
530 CONTINUE
LING = LING + NUOP2
530 CONTINUE
IF (MPRZ,EG,0) GO TO 537
IF (L,EG,1) WRITE (6,5) NSTRIP
DO 535 JM = 1,NDOP
DO 535 JN = 1,NDOP
WRITE (6,15) K, KB, JM, (CH1JM,JMS,KB), JMS-1,NDOP(1)
535 CONTINUE
537 CONTINUE
540 CONTINUE
RETURN
END

```

01/15/71

```

AIC - EFN SOURCE STATEMENT - (FMIS) -
SUMMARY AIC(S),S,HR,MF,MEL,JOBN,ICDO,VEL,NGUST,MPRZ)
CONTINUE (G,0) GO TO 530
IF (MUST,STRIP) GO TO 530
DO 520 I1 = 1,NDOP1
G(I1) = 0.
DO 520 L = 1,NDOP2
G(L) = LING * L
G(1) = 50.11443E-04*VEL*0(11),L1)PCGUST(LG1+G(I1))
530 CONTINUE
LING = LING + NUOP2
530 CONTINUE
IF (MPRZ,EG,0) GO TO 537
IF (L,EG,1) WRITE (6,5) NSTRIP
DO 535 JM = 1,NDOP
DO 535 JN = 1,NDOP
WRITE (6,15) K, KB, JM, (CH1JM,JMS,KB), JMS-1,NDOP(1)
535 CONTINUE
537 CONTINUE
540 CONTINUE
RETURN
END

```


~~UNCLASSIFIED~~
Security Classification

DOCUMENT CONTROL DATA - R & D		
<i>(Security classification of title, body of abstract and indexing annotation must be entered when the overall report is classified)</i>		
1. ORIGINATING ACTIVITY (Corporate author) Douglas Aircraft Company Aircraft Division		2a. REPORT SECURITY CLASSIFICATION UNCLASSIFIED
		2b. GROUP
3. REPORT TITLE Subsonic Unsteady Aerodynamics for General Configurations Part I, Vol II - Computer Program H7WC		
4. DESCRIPTIVE NOTES (Type of report and inclusive dates) Final Technical Report, December 1969 - September 1970		
5. AUTHOR(S) (First name, middle initial, last name) J. P. Giesing T. P. Kalman W. P. Rodden		
6. REPORT DATE November 1971	7a. TOTAL NO. OF PAGES 143	7b. NO. OF REFS 0
8a. CONTRACT OR GRANT NO. F33615-70-C-1167	9a. ORIGINATOR'S REPORT NUMBER(S) AFFDL-TR-71-5, Part I, Vol II	
b. PROJECT NO. 1370	9b. OTHER REPORT NO(S) (Any other numbers that may be assigned this report) MDC-J0944	
c. Task No 137003		
d.		
10. DISTRIBUTION STATEMENT Distribution limited to U. S. Government agencies only; test and evaluation; statement applied 23 June 1971. Other requests for this document must be referred to Air Force Flight Dynamics Laboratory/FY, Wright-Patterson AFB, Ohio 45433.		
11. SUPPLEMENTARY NOTES	12. SPONSORING MILITARY ACTIVITY Air Force Flight Dynamics Laboratory Wright-Patterson AFB, Ohio 45433	
13. ABSTRACT <p>Two methods of accounting for body-lifting surface interference in unsteady flow are considered. The first method is described in Part I of this report, while the second will be described in Part II to follow.</p> <p>The first method is a direct application of nonplanar lifting surface elements to both the lifting surfaces and the body surfaces. The body is treated as an annular wing. This type of idealization must be used with an axial doublet introduced to account for body incidence effects. The undesirable effects of the annular wing representation are then reduced.</p> <p>The second approach, to be described in Part II, uses an image system and an axial singularity system to account for the effects of the bodies.</p> <p>This report also describes an improvement of the Doublet-Lattice Method of Albano and Rodden. The improvement pertains to wing-tail problems where there exists a small vertical (non-zero) separation between the wing and tail planes. Such problems can now be handled with ease.</p>		

DD FORM 1 NOV 68 1473

~~UNCLASSIFIED~~
Security Classification

14. KEY WORDS	LINK A		LINK B		LINK C	
	ROLE	WT	ROLE	WT	ROLE	WT
Unsteady Aerodynamics						
Subsonic Flow						
Aerodynamic Interference						